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AFFDL-TR-71-5
PART I, VOL II

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**SUBSONIC UNSTEADY AERODYNAMICS
FOR GENERAL CONFIGURATIONS**

PART I, VOL II—COMPUTER PROGRAM H7WC

*J. P. GIESING
T. P. KALMAN
W. P. RODDEN*

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TECHNICAL REPORT AFFDL-TR-71-5, PART I, VOL II

NOVEMBER 1971

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SUBSONIC UNSTEADY AERODYNAMICS FOR GENERAL CONFIGURATIONS

PART I, VOL II—COMPUTER PROGRAM H7WC

J. P. GIESING
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FOREWORD

This report was prepared by the Douglas Aircraft Company, Long Beach, California, for the Aerospace Dynamics Branch, Vehicle Dynamics Division, Air Force Flight Dynamics Laboratory, Wright-Patterson Air Force Base, Ohio, under Contract F33615-70-C-1167. This research was conducted under Project 1370, "Dynamic Problems in Military Flight Vehicles," and Task 137003, "Prevention of Dynamic Aeroelastic Instabilities in Advanced Military Aircraft." Mr. S. J. Pollock of the Aerospace Dynamics Branch was Task Engineer.

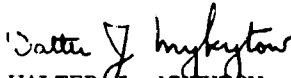
This report consists of two parts with two volumes for each part. This volume, Volume II of Part I is the Computer Program H7WC. Volume I of Part I consists of the method of direct application of nonplanar lifting surface elements. Volume I of Part II, contains a method which uses an image system and an axial singularity system to account for the effects of the bodies and Volume II of Part II is the Computer Program N5KA.

The work reported herein was conducted during the period of December 1969 to September 1970.

The Principal Investigator was Joseph P. Giesing. Mrs. Terez P. Kalman was responsible for implementing the method on the computer. Donald H. Larson aided in this implementation and acted as consultant for computer problems. Dr. William P. Rodden, a McDonnell Douglas Company Consultant, contributed many valuable ideas to the project. Others have made significant contributions to this project including Messrs. D. S. Warren and W. E. Henry.

The report was submitted by the authors in November 1970 for publication as an AFFDL Technical Report.

This technical report has been reviewed and is approved.


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A B S T R A C T

Two methods of accounting for body-lifting surface interference in unsteady flow are considered. The first method is described in Part I of this report, while the second will be described in Part II to follow.

The first method is a direct application of nonplanar lifting surface elements to both the lifting surfaces and the body surfaces. The body is treated as an annular wing. This type of idealization must be used with an axial doublet introduced to account for body incidence effects. The undesirable effects of the annular wing representation are then reduced.

The second approach, to be described in Part II, uses an image system and an axial singularity system to account for the effects of the bodies.

This volume contains the computer program and contains the FORTRAN listing. Volume I contains the development of the theory, correlation of theory with experimental data, and the parametric study.

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NOMENCLATURE

A	Reference total area
[A]	Matrix of box areas
a_{inm}	Polynomial mode coefficients for mode i
AIC	Matrix of influence coefficients relating generalized forces to generalized deflections for submodes
c	Chord length
\bar{c}	Reference chord length
C_m	Pitching moment coefficient $\left(\frac{\text{moment}}{q A \bar{c}} \right)$, (+ nose up)
C_n	Yawing moment coefficient $\left(\frac{\text{moment}}{q A s} \right)$ (+ nose right)
C_l	Rolling moment coefficient $\left(\frac{\text{moment}}{q A s} \right)$ (+ clockwise)
C_Y	Force coefficient in y-direction $\left(\frac{\text{Force}}{q A} \right)$ (+ out right wing)
C_Z	Force coefficient in z-direction $\left(\frac{\text{Force}}{q A} \right)$ (+ vertically up)
c_m	Local moment coefficient $\left(\frac{\text{moment}}{q c^2} \right)$
c_n	Local normal force coefficient $\left(\frac{\text{normal force}}{qc} \right)$
e	Strip semi-width
ϵ	Normalized deflection normal to surface (h/s)
h	Deflection normal to surface
K	Kernel
k_r	Reduced frequency, $\omega \bar{c} / 2U_\infty$
L	Lift
M	Mach number; also moment
Q_{ij}	Generalized force
q	Dynamic pressure
\bar{q}	Generalized coordinates
R_o	Body radius
r	$\sqrt{(y-\eta)^2 + (z-\xi)^2}$

s	Semi-span
U_{∞}	Free-stream velocity
W	Unnormalized normalwash
w	W/U_{∞}
w_r	$w - w_b$
x, y, z	Coordinates of receiving points
$\bar{x}, \bar{y}, \bar{z}$	Element coordinates of receiving points
x_0	Gust reference axis
α	Angle of attack; also a function defined by Equation 2.1-15
β	$\sqrt{1 - M^2}$; also control surface deflection
Γ_g	Gust dihedral angle. ($\Gamma_g = 0$ if gust velocity is vertical)
γ	Dihedral angle
ΔC_p	Lifting pressure coefficient $\frac{P_{\text{lower}} - P_{\text{upper}}}{q}$
δ	Symmetry index (right and left symmetry); also tab deflection
ϵ	Ground effect index
λ	Wave length
ξ, η, ζ	Coordinates of sending points
$\bar{\xi}, \bar{\eta}, \bar{\zeta}$	Coordinates of sending points in element coordinates
ρ_2	Density at sea level
σ	Lateral coordinate in the plane of the surface
ω	Frequency

Subscripts and Superscripts

a	body axis
B	body
c	center
f	Body or fuselage
g	gust

i	Deflection mode
j	Pressure mode
L. E.	Leading edge
LL	Lower left
LR	Lower right
R	Axis about which moments are taken
r	Receiving
s	Sending
UL	Upper left
UR	Upper right
y	y-direction
z	z-direction
1/4	One quarter chord point of a lifting surface box
3/4	Three quarters chord point of a lifting surface box

1.0 INTRODUCTION

Computer program H7WC is the result of modifying and extending program H7WB. Program H7WB generates aerodynamic influence coefficients and gust loads for nonplanar surfaces. The modifications to this program include:

- (1) An extension to accept polynomial mode input. (Subroutine GENQ)
- (2) The addition of a capability to determine generalized forces. (Subroutine GENF)
- (3) The addition of a capability to account for body angle of attack and camber effects using an axial pressure doublet system. (Subroutines AUGW, TKER)
- (4) A refined version of the Doublet-Lattice Method to properly handle problems involving small vertical gaps between wing and tail surfaces. (Subroutine IDF2)
- (5) The addition of the Quasi-Inverse for solving large numbers of modes or submodes, and for solving future modes with a minimum of computing effort. (Subroutines QUAS, FUTSOL, CXSS)

Program H7WC can handle almost any type of configuration if it is idealized properly. The axial pressure doublets give the proper slender body flow field for an axially symmetric body of any shape. The body/lifting-surface interference is taken care of using lifting surface panels on the body surface. The interference elements must lie on an idealized body of constant cross sectional shape. The rules that apply to the placement of a grid of boxes on the aircraft lifting surfaces also apply to the interference surfaces on the bodies. These rules are:

- (1) Boxes (trapezoidal elements) are arranged in strips parallel to the free-stream. The aspect ratio of such boxes should not be large. For the unsteady case, an aspect ratio of order unity or less is preferred.

- (2) Surface intersections, surface edges, fold lines, and hinge lines should lie on box boundaries.
- (3) For wing-tail type configurations the strip boundaries on the tail must be aligned with those on the wing. Strips should be concentrated near the wing tips, near control surface edges and in any region where the span loading changes rapidly.
- (4) The box length must be small compared to the basic wave length.

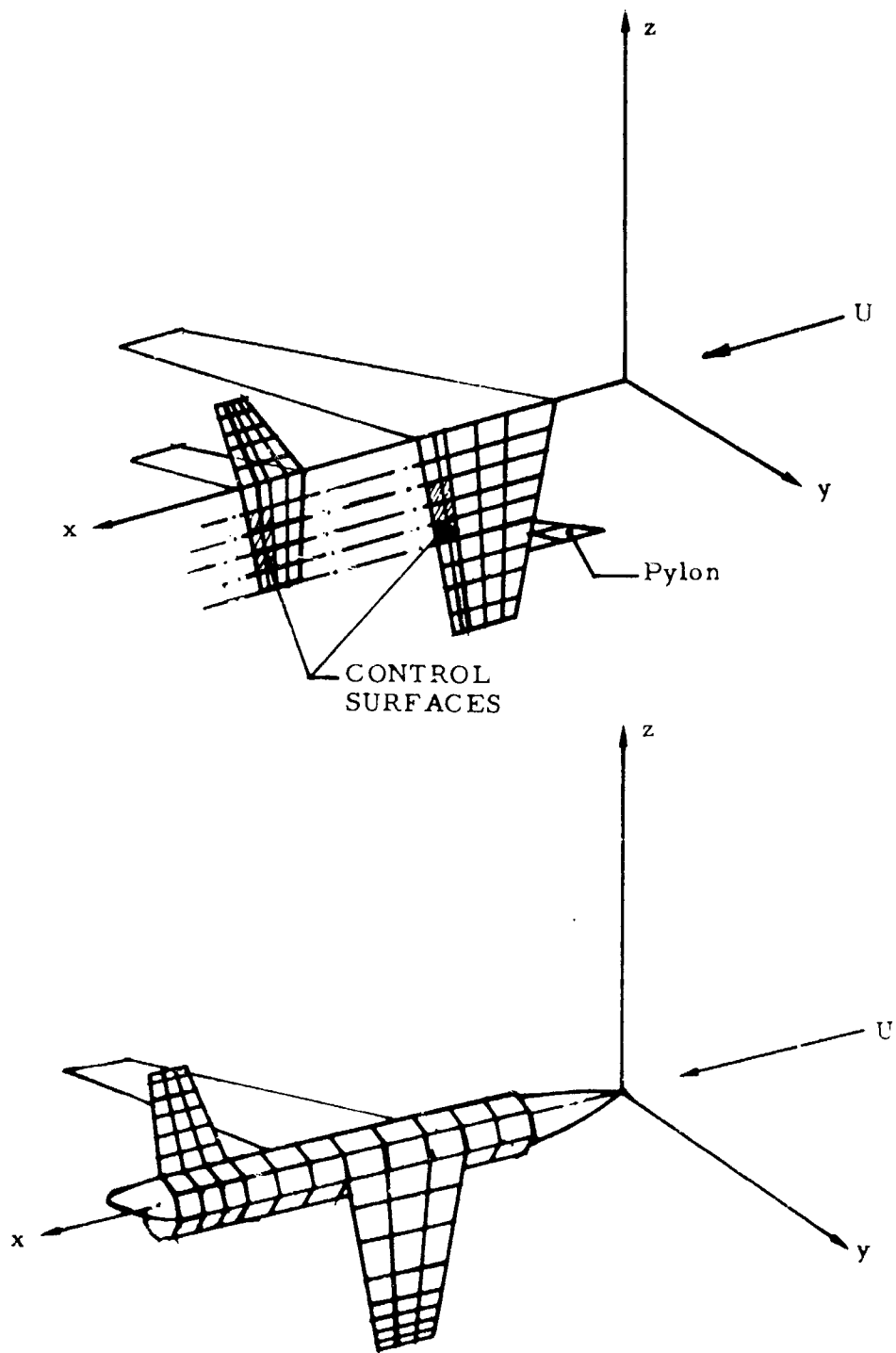
$$\frac{\Delta x}{\lambda} \leq 0.04$$

Boxes should also be concentrated near control surface hinge-lines and in all regions where the upwash boundary conditions are discontinuous.
- (5) The body cross-sectional shape on which interference lifting surfaces are placed must be constant in the x-direction even for wing-tail problems. (This does not apply to the slender body solution for which the radius distribution is used.)

Sketch 1,1-1 gives examples of these 5 rules.

Because of the numerical techniques involved, the two-dimensional radial distance, $\sqrt{(y-\eta_e)^2 + (z-\zeta_e)^2}$, between a control point, y, z , (3/4-chord point of a box) and a box edge, η_e, ζ_e , may never approach zero. It does not matter whether the control point is on the same or on a different surface or whether it is up-or downstream from the box in question. This is easily understood for a control point in the wake of the box since the control point may not approach the bound leg of the horseshoe vortex. However, it is also true upstream due to the approach taken for the unsteady effects. (If $k_r = 0$ the upstream effect at $\sqrt{(y-\eta)^2 + (z-\zeta)^2} \rightarrow 0$ vanishes).

Wing-body-tail problems must be handled using only one interference cross-section. That is, one constant cross-sectional shape must be used both near the wing and near the tail. If this rule is not followed the wakes of the lifting surface element lying on the cross-section near the wing may impinge on the tail surface.



Sketch 1.1-1

When a body mode is input it must be input both to the axial elements and to the interference body panels. It must be remembered, however, that the body is in motion either in the z-or y-directions while the lifting surfaces on the body are in motion normal to themselves. Thus the mode input for a particular body interference panel must be multiplied by $\cos\gamma$ for (+) z-motion and $-\sin\gamma$ for (+) y-motion where γ is the dihedral angle of the panel.

The following list gives the program limits.

1. The maximum number of unknowns, i. e., the aerodynamic boxes on the lifting surfaces plus the number of body elements, is 400.
2. The maximum number of aerodynamic degrees of freedom, in case of AIC calculations, is 130.
3. The maximum number of degrees of freedom per strip in case of AIC calculations is restricted to 4.*
4. The maximum number of modal right-hand sides is 40 per case.
5. The maximum number of aerodynamic boxes per spanwise division (strip) is 30.
6. The maximum number of spanwise divisions (strips) is 50.
7. The maximum number of bodies is 20.

Program H7WC was used successfully on the following computers: GE635, IBM 7094 and IBM 360/65. Core requirement for the program on the various computers is as follows:

<u>GE635</u>	<u>IBM 7094</u>	<u>IBM 360/65</u>	<u>CDC 6600</u>
28K words	32K words	126K bytes	100K bytes

If greater core capacity is available it is desirable to use it up to 260K bytes. The CDC version is slightly modified in dimensions.

* This restriction comes from operational requirements on the IBM 7094. The number of degrees of freedom can easily be increased to the 7 discussed in Appendix D in Part I, Vol. I by increasing the dimension 4 to 7 in subroutines AIC, AERO, and FINAL.

2.0 COMPUTER PROGRAM H7WC

2.1 Input Procedure

2.1.1 Input Sheets

The input sheets for program H7WC are shown on the next three pages. The first three cards represent general case data that is input once per case. The next four* cards (4, 5, 6, 7) represent panel data that is repeated per panel. The panel data must be input in a certain order. The panel data for all "regular" lifting surfaces, (wing, tail, etc.) are input first. The panel data for all interference lifting surfaces (used for the bodies) is input last.

If a body can oscillate in both the z- and y-directions (nacelle) then it must be replaced by two bodies: one that moves in the z-direction and one that moves in the y-direction. The subsequent three** cards (8, 9, 10) are body data necessary for the slender body analysis. This data is repeated per body. Card 11 is used for miscellaneous modal and AIC data and must be completed in either case. Card 12 is used when a gust is desired. Card 13 indicates a series of cards that are used to identify lifting surface boxes with strips. Card 14 indicates one or more cards that identify the degrees of freedom to be used for each panel. This card is necessary only if AIC's are desired.

Cards 15 and 16 are used to identify axes for the AIC degrees of freedom. Card 17 is used for modal control data. Card 18 is used to input control data for panel modes, while card 19 represents a series of cards, repeated as necessary, to indicate polynomial mode coefficients for panels. Cards 20 and 21 are similar to cards 18, 19 except the modes pertain to the z and y modes of the bodies.

* There may be more than four cards depending on the number necessary to present all θ_i and τ_i values.

* There may be more than three cards depending on the number necessary to present F and RAD.

DATE _____

53	54	55	56	57	58
----	----	----	----	----	----

H	7	W	C
73	74	75	76

PUNCH IN ALL CARDS

PROGRAM NO.

[illegible]

DIRECTIONS FOR REVISION

DO NOT PUNCH BLANK COLUMNS

NO UNDERPUNCHES IN SIGN FIELDS.

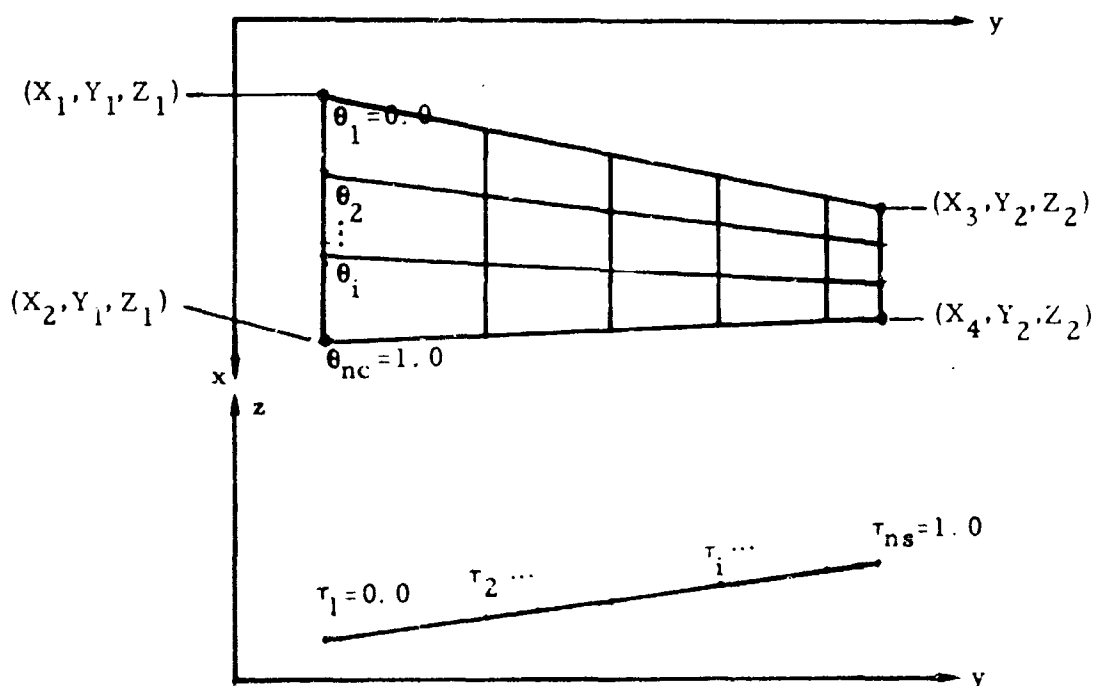
REPEAT PER PANEL REPEAT PER BODY

2. 1. 2 Description of Input Data

<u>Item No.</u>	<u>Description</u>
1	The header card is used to describe the current run being made. Any alphanumeric information can be used, placed in cc 1-60. The first 12 card columns should contain the tape title for the AIC's, in the case when AIC's are generated and are to be saved on master tape for later use.
2	<u>M, FMACH</u> - Mach number, usual definition.
3	<u>A, ACAP</u> - Reference area; usually total area of wing.
4	<u>c̄, REFCHD</u> - Reference chord, usually average chord of wing
5	<u>s, REFSPN</u> - Reference semispan.
6	<u>δ, NDELT</u> - Symmetry flag ($\delta = 1$ sym., $\delta = -1$, antisym. $\delta = 0$, no sym.).
7	<u>NP, NOPAN</u> - Total number of panels on all lifting surfaces.
8	<u>NB</u> - Total number of bodies.
9	<u>NRF</u> - Number of reduced frequencies for which the analysis has to be performed (repeated) - recommended maximum is 6 per case.
10	<u>NCORE</u> - Size of total solution, i.e., $NCORE = N \times M$ where N is the total number of elements or unknowns (including z and y fuselage elements) and M is the number of modes or the order of the AIC matrix if obtained.
11	<u>N!</u> - If "1", program assumes that case has been run previously and that a quasi-inverse exists on a tape (identified by NTPI) and only new modes are being run. If "0", program runs a new case.

Item No.Description

- 12 N2 - If "0", polynomial modes are to be input. If "1" AIC's are to be calculated.
- 13 N3 - Data flag. If "1", matrix is printed; if "0", it is not printed.
- 14 N4 - Detail print flag for slender body solution; 1 to print, 0 otherwise.
- 15 KR - Array of reduced frequency, $KR = \frac{\omega \bar{c}}{2 U_\infty}$.
- 16,17,18 X1 X2 X3 - Panel edge coordinates.
- 19,20,21 X4 Y1 Y2 X1, Y1, Z1 (Inboard leading edge)
- 22,23 Z1 Z2 X2, Y1, Z1 (Inboard trailing edge)
- X3, Y2, Z2 (Outboard leading edge)
- X4, Y2, Z2 (Outboard trailing edge)
- 24 NS - Number of spanwise divisions for panel considered.
- 25 NC - Number of chordwise divisions for panel considered.



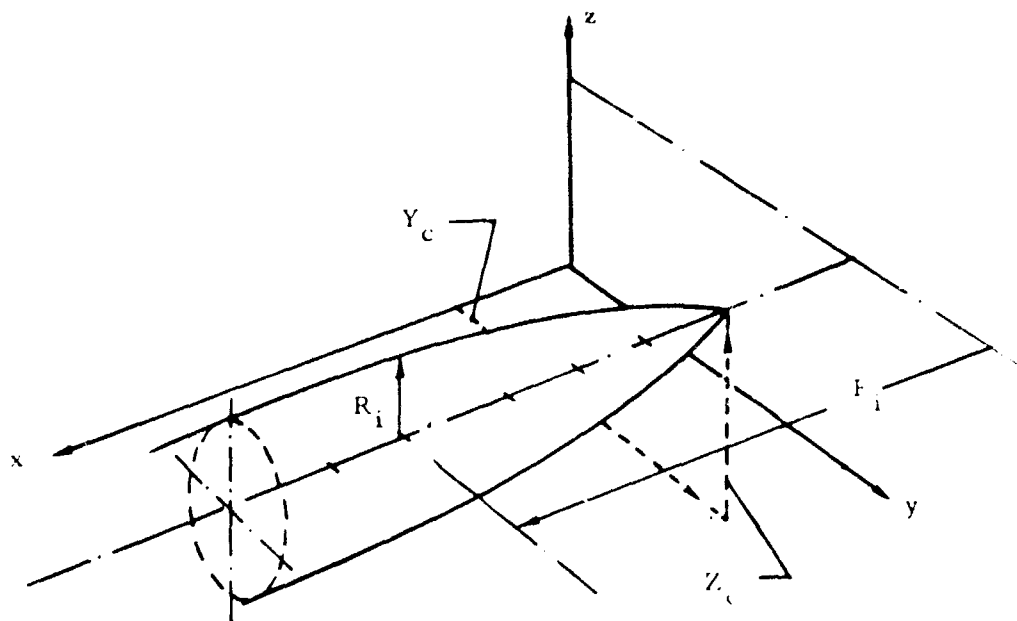
SKETCH 2.1.2-1

<u>Item No.</u>	<u>Description</u>
26	<u>COEF</u> - Scale factor for panel deflection modes. If no scale factor is desired, must be set to 1.0.
27	<u>θ_i</u> - Chordwise divisions in fraction of chord. Usually varies from 0 at leading edge to 1.0 at the trailing edge.
28	<u>τ_i</u> - Spanwise divisions in fraction of panel span. Usually 0.0 at inboard chord and 1.0 at outboard chord.

Repeat all panel data per panel (Items 16 through 28) Panels are input in the following sequence: wing, tail, etc., panels, then body interference panels.

29	<u>Z_c</u> - Z-coordinate of body axis.
30	<u>Y_c</u> - Y-coordinate of body axis.
31	<u>NF</u> - Number of divisions on body.
32	<u>NZ</u> - If "1", the program will allow for doublets in the Z-direction and thus will be able to account for Z-or upwash. If not, set NZ = 0.
33	<u>NY</u> - If "1", program will allow for Y-doublets and thus Y-or sidewash.
34	<u>COEF</u> - Scale factor for body deflection mode.
35	<u>MK1</u> - Each body has associated with it interference lifting surfaces. MK1 represents the first box on these surfaces.
36	<u>MK2</u> - The last box on the body interference lifting surface panels. Interference lifting surface panels are input after the regular lifting surfaces are input.
37	<u>E</u> - Endpoints of body divisions (starts at leading edge and ends at trailing edge). E x coordinates of body element end points.
38	<u>RAD</u> - Radius at endpoints.

Repeat all body data per body (Items 29 through 38)



SKETCH 2.1.2-2

Item No.	Description
39	<u>NSTRIP</u> - Total number of strips on all lifting surfaces.
40	<u>NB1</u> - Sequence number of first strip. Omit if no AIC's are desired.
41	<u>NB2</u> - Sequence number of last strip. Omit if no AIC's are desired.
42	<u>NPR1</u> - Print flag. If "1", print all AIC solutions. Otherwise use "0" (usually = 0).
43	<u>NPR2</u> - Print flag. If "1", print AIC's. Otherwise use "0" (usually = 0).
44	<u>NPR3</u> - Stability derivative flag. If "1", print static stability derivatives. If "3", print static-and-dynamic stability derivatives. Otherwise use "0". Note that, if NPR3 = 3, NRF must be ≥ 2 and FR(1) must be 0.0. Usual setting is NPR3=1.
45	<u>JOBNO</u> - AIC tape identification. Ignore if no AIC's.
46	<u>JOBNO2</u> - Not used.
47	<u>NGUST</u> - Gust flag. Use "1" for gust, "0" otherwise.

<u>Item No.</u>	<u>Description</u>
48	<p><u>JSP</u> - Second symmetry plane (the plane $Z = 0$).</p> <p>-1 ground effect (antisymmetric)</p> <p>1 biplane or jet (symmetric)</p> <p>0 no symmetry</p>
49	<p><u>NPC</u> - Mode selector for AIC generation.</p> <p>Ignore if no AIC's. Use "0" if alternative #1 is used i. e. plunging, pitching, control surface rotation and tab rotation. Use "1" if alternative #2 is used, i. e., three cambering modes plus control surface and tab rotation.</p>
50	<p><u>NSV</u> - Total number of strips on all vertical panels which lie on the symmetry plane $Y = 0$.</p> <p>Vertical panels lying on $Y = 0$ must be input before the other panels. For example the vertical portion of the T-tail must be input first.</p>
51	<p><u>NBV</u> - Total number of boxes on vertical panels.</p>
52	<p><u>NYAW</u> - Stability derivative override flag. If "1", yaw coefficients will be calculated, if "0", pitch coefficients will be calculated.</p> <p>If $NDEL T = 1$, $NYAW = 0$</p> <p>$NDEL T = -1$, $NYAW = 1$</p> <p>$NDEL T = 0$, $NYAW = 0$ or 1</p>
53	<p><u>GZRO</u> - Gust reference plane dihedral in degrees. For 0 dihedral ($GZRO = 0$) the gust velocities are vertical (in the Z-direction). Omit if gust is not desired.</p>
54	<p><u>XZRO</u> - Gust reference point. Point at which sinusoidal gust velocity is unity. Omit if gust is not desired.</p>

<u>Item No.</u>	<u>Description</u>
55	<u>VEL</u> - Aircraft velocity in. /sec. Omit if gust is not desired.
56	<u>WG</u> - Gust vertical velocity at XZRO, in. /sec. Omit if gust is not desired.
<hr/>	
Omit the gust card if the gust is not desired. (Items 53 through 56)	
<hr/>	
57	<u>LIM</u> - Summation limits for chordwise integration. Usually LIM1 is the first box in a strip and LIM2 is the last box where all boxes are numbered consecutively.
58	<u>NOP</u> - Panel number.
59, 60, 61 62, 63, 64 65	IS1 IS2 IS3 - AIC degree-of-freedom selector. If alternative #1 is used (i. e. if NPC = 0) then IS1 through IS6 (IS7 not used) represent: plunging, pitching, control surface rotation, tab rotation, control surface plunging, tab plunging, respectively. If Alternative #2 is used then IS1 through IS7 represent: three cambering modes on the main surface, control surface rotation, tab rotation, control surface plunging, and tab plunging, respectively.
66	<u>XHI</u> - Reference axes measured in fractions of local chord on the inboard edge of a panel. (Further description following XHO.) Ignore if AIC's are not needed.
67	<u>XHO</u> - Reference axes measured in fractions of local chord on the outboard edge of a panel. For Alternative #1 5 quantities are needed: the elastic axis, the leading edge of the control surface, the rotation point of the control surface, the tab leading edge and rotation point, respectively. The data pair XHI and XHO are repeated for each panel.

Item No.Description

Omit data Items #58 through #67 if AIC's are not required.

- 68 NM, NMD - Total number of modes, i.e., max NQ.
- 69 NT, NTA - Total number of ARQ values.
- 70 N5 - If "1", save the quasi-inverse on tape so that future modes may be considered. Otherwise set to "0".
- 71 N6 - Type of mode input flag
 0 - polynomial
 1 - input h, dh/dx from tape
 2 - input h, dh/dx from cards
- 72 N7 - If "1", calculate pressure forces and moments.
 Use "0" otherwise.
- 73 NTPI - Quasi Inverse tape identification. Written on tape when Quasi Inverse is formed and used to identify the Quasi Inverse when it is retrieved for additional mode cases.
- 74 NTPM - Identification of input mode (h, dh/dx) tape.
-

The input to follow is associated with the input modes. Equations will be written to define the variables. For panels (lifting surfaces) the deflections will be normal to the surface while for bodies the deflections will be in the Z- (if any) and Y- (if any) directions. For the "NR"th panel or body the deflections in the "NQ"th mode are calculated as follows:

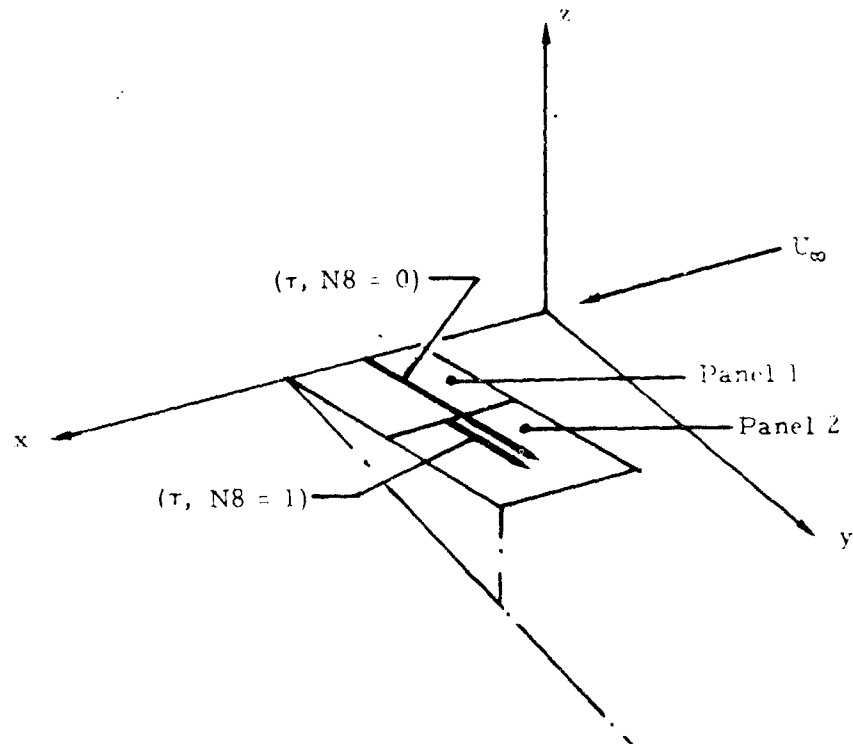
$$(f) \begin{matrix} (NR) \\ NQ \end{matrix} = \text{COEF}^{(NR)} \sum_{N=0,1,\dots} \sum_{M=0,1,\dots} \left(\frac{x}{s}\right)^N \left(\frac{r}{s}\right)^M \text{ARQ}_{(NR, NQ, N, M)}$$

where r is in a radial direction. The origin of the radius is either at the

origin of coordinates, when $N8 = 0$, or at the inboard edge (for the panel) or the axis (for a body) if $N8$ is set to 1.*

$$\tau = \sqrt{(y - (N8)Y_1^{(NR)})^2 + (z - (N8)Z_1^{(NR)})^2} \quad \text{for panels}$$

$$\tau = \sqrt{(y - (N8)Y_c^{(NR)})^2 + (z - (N8)Z_c^{(NR)})^2} \quad \text{for bodies}$$



SKETCH 2.1.2-3

* Since each panel is planar, τ is a spanwise distance in the plane measured from its inboard edge or from the origin $y = z = 0$; τ is a radial distance only in the sense that each panel may have a different dihedral. The use of this radial distance is not meant to imply that the panels are curved.

<u>Item No.</u>	<u>Description</u>
75	<u>MP, NMTP</u> - Total number of ARQ coefficients for panels only.
76	<u>MB, NMTB</u> - Total number of ARQ coefficients for bodies.
77	<u>NR</u> - Panel number and/or body number.
78	<u>NQ</u> - Mode number.
79	<u>NA, NARQ</u> - Number of terms (ARQ) in mode NQ, panel and/or body NR.
80	<u>N8</u> - Flag that sets origin of radial variable . If "0", the x-axis is used. If "1", inboard edge or axis of body is used as origin (see equation for τ).

The four items NR, NQ, NA, N8 are repeated for each panel or body until all panels and bodies are mentioned. Even if a panel or body is not in motion (NA = 0) the set of four items must be given.

The direction of motion of a body is given by the NZ or NY flags. Thus if body pitching is given and NY is flagged, then the body will be given a yaw. If a body can move in both the Z and Y directions (a nacelle) then it must be replaced with two bodies: one with NZ = 1 and one with NY = 1.

81	<u>N, LARQ(1)</u> - Power of x/s in the mode polynomial.
82	<u>M, LARQ(2)</u> - Power of τ/s in the mode polynomial.
83	<u>ARQ</u> - Coefficient of $(x/s)^N (\tau/s)^M$ in the mode polynomial.

Omit modal data cards if modes are not desired (Items 68 through 83).

If modes are input using the options N6=1, or N6=2, omit Items 73 through 83, and add the following data items:

<u>Item No.</u>		
84	<u>NBOX</u>	- Total number of boxes on all panels.
85	<u>B</u>	- Generalized force integration matrix (box area) X (displacement 1/4-chord of box).
86	<u>H</u>	- Displacement of 3/4-chord point normalized by semispan, h/s.
87	<u>DH1</u>	- Derivative of normalized displacement by (x/s) at 3/4-chord point of box. $d(h/s)/d(x/s)$.
88	<u>NBEL</u>	- Total number of body elements.
89	<u>B</u>	- Generalized force integration matrix (body element length) X (diameter) X (displacement 1/2 -chord of element).
90	<u>H</u>	- Displacement of 1/2-chord point normalized by semispan, h/s.
91	<u>DH1</u>	- Derivative of normalized displacement by (x/s) at 1/2-chord point of element. $d(h/s)/d(x/s)$.
92	<u>DH2</u>	- $d^2(h/s)/d(x/s)^2$.

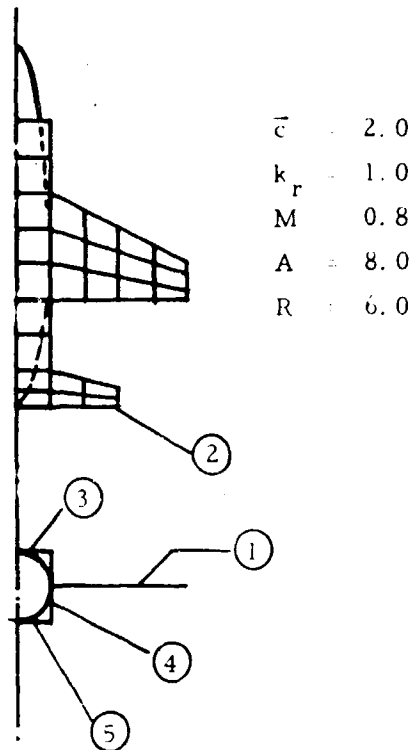
NOTE: If N6=1, Items #84 through #92 are input from cards according to the formats shown on page 8.

If N6=2, Items #84 through #92 are read from tape unit MTAPE, as one record per variable, NBOX, (B_i , $i = 1, NBOX$), (H_i , $i = 1, NBOX$) and ($DH1_i$, $i = 1, NBOX$). If bodies are also present, five more variables are read from tape unit MTAPE; again, as one record per variable: NBEL, (B_i , $i = 1, NBEL$), (H_i , $i = 1, NBEL$), ($DH1_i$, $i = 1, NBEL$) and ($DH2_i$, $i = 1, NBEL$).

For each additional mode, repeat Items #85 through #92.

2.2 Example Case

The following case is to be viewed only as an example of the proper input procedure and not as an example of an ideal or optimum idealization. The configuration is shown in Sketch 2.2-1.



Sketch 2.2-1

The fuselage is divided up into 10 equal elements for the slender body calculation. The divisions along with the radius is given as follows:

$$F_k = 0, 1.0, 2.0, 3.0, 4.0, 5.0, 6.0, 7.0, 8.0, 9.0, 10.0$$

$$RAD_k = 0, 0.7, 0.8, 0.9, 0.95, 1.0, 0.9, 0.8, 0.7, 0.5, 0.0$$

The axis of the fuselage lies on the x-axis, $Z_c = Y_c = 0$ and it is assumed that the fuselage moves only in the z-direction ($NZ = 1$). The modes of

oscillation are symmetric ($\delta = 1.0$) pitching about the middle of the fuselage and plunging

$$h/s = -5.0 + 5 x/s \text{ (where } s = 5)$$

$$h/s = 1.0$$

These modes apply to the entire configuration. The panels are numbered from 1 to 5 in Sketch 2.2-1.

The panels are given as follows:

- (1) $X_1 = 4.0, X_2 = 7.0, X_3 = 6.0, X_4 = 7.0$
 $Y_1 = 1.0, Y_2 = 5.0, Z_1 = 0, Z_2 = 0$
 $\theta = 0, 0.333, 0.666, 1.0, \tau = 0, 0.25, 0.50, 0.75, 1.0$
- (2) $X_1 = 9, X_2 = 10, X_3 = 9.5, X_4 = 10.0$
 $Y_1 = 1.0, Y_2 = 3.0, Z_1 = 0, Z_2 = 0$
 $\theta = 0, 0.5, 1.0$
 $\tau = 0, 0.5, 1.0$
- (3) $X_1 = 2.0, X_2 = 10.0, X_3 = 2.0, X_4 = 10.0$
 $Y_1 = 0, Y_2 = 1.0, Z_1 = 1.0, Z_2 = 1.0$
 $\theta = 0, 0.125, 0.25, 0.375, 0.50, 0.625, 0.75, 0.875, 0.9375, 1.0$
 $\tau = 0, 1.0$
- (4) $X_1, X_2, X_3, X_4, \theta$ same as (3)
 $Y_1 = 1.0, Y_2 = 1.0, Z_1 = 1.0, Z_2 = -1.0$
 $\tau = 0, 0.5, 1.0$
- (5) $X_1, X_2, X_3, X_4, \theta, \tau$ same as (3)
 $Y_1 = 1.0, Y_2 = 0.0, Z_1 = -1, Z_2 = -1$

The input sheets are shown on the next several pages. The output is shown on the subsequent pages.

2.2.1 Input Sheets

2.2.2 Output

WING TEST CASE WING-BODY-TAIL

00 ARRAY OF REDUCED FREQUENCIES 00

1.000000

REP. CMOHD = 2.00000 REP. SEMI-SPAN = 5.00000 REP. AREA = 16.0
MACH NO. = 0.80000 BETA = 0.80000

000 PANEL NO. 1 INPUT VALUES 000

X1 = 4.000000 X2 = 7.000000 Y1 = 1.000000 Z1 = 0.0
X3 = 6.000000 X4 = 7.000000 Y2 = 1.000000 Z2 = 0.0
NC = 4 NS = 5 NOELT = 1 NO. OF PANELS = 5

4 CIRCUMFERENCE DIVISIONS FOR PANEL 1

0.0 0.33332990E 00 0.66666980E 00 0.10000000E 01

5 SPANWISE DIVISIONS FOR PANEL 1

0.0 0.25000000E 00 0.50000000E 00 0.75000000E 00 0.10000000E 01
20 000 XI ELEMENTS FOR PANEL 1000
0.40000000E 01 0.40000000E 01 0.60000000E 01 0.70000000E 01 0.45000000E 01 0.53333244E 01
0.61666670E 01 0.70000000E 01 0.90000000E 01 0.56666594E 01 0.63333340E 01 0.70000000E 01
0.95000000E 01 0.59999943E 01 0.75000000E 01 0.70000000E 01 0.60000000E 01 0.63333292E 01
0.66666670E 01 0.70000000E 01

20 000 ETA ELEMENTS FOR PANEL 1000

0.10000000E 01 0.10000000E 01 0.10000000E 01 0.10000000E 01 0.70000000E 01 0.20000000E 01
0.20000000E 01 0.20000000E 01 0.30000000E 01 0.30000000E 01 0.30000000E 01 0.30000000E 01
0.40000000E 01 0.40000000E 01 0.40000000E 01 0.40000000E 01 0.50000000E 01 0.50000000E 01
0.50000000E 01 0.50000000E 01

20 ZEE ELEMENTS FOR PANEL NO. 1

0.0 0.0 0.0 0.0 0.0 0.0
0.0 0.0 0.0 0.0 0.0 0.0
0.0 0.0 0.0 0.0 0.0 0.0
0.0 0.0

4 000 XI-J ELEMENTS FOR PANEL 1000

0.42500000E 01 0.47500000E 01 0.52500000E 01 0.57500000E 01

C.WIGGLES FOR PANEL NO. 1 00000

0.27500000E 01 0.22499990E 01 0.17500000E 01 0.12500000E 01

000 PANEL NO. 2 INPUT VALUES 000

X1 = 9.000000 X2 = 10.000000 Y1 = 1.000000 Z1 = 0.0
X3 = 9.500000 X4 = 10.000000 Y2 = 1.000000 Z2 = 0.0
NC = 5 NS = 5 NOELT = 1 NO. OF PANELS = 5

3 CIRCUMFERENCE DIVISIONS FOR PANEL 2

0.0 0.50000000E 00 0.10000000E 01

3 SPANWISE DIVISIONS FOR PANEL 2

0.0 0.50000000E 00 0.10000000E 01
9 000 XI ELEMENTS FOR PANEL 2000
0.40000000E 01 0.45000000E 01 0.10000000E 02 0.92500000E 01 0.46250000E 01 0.10000000E 02
0.95000000E 01 0.97500000E 01 0.10000000E 02
9 000 ETA ELEMENTS FOR PANEL 2000
0.10000000E 01 0.10000000E 01 0.10000000E 01 0.20000000E 01 0.20000000E 01 0.20000000E 01
0.30000000E 01 0.30000000E 01 0.30000000E 01
9 ZEE ELEMENTS FOR PANEL NO. 2

0.0 0.0 0.0 0.0 0.0 0.0
0.0 0.0 0.0

2 000 XI-J ELEMENTS FOR PANEL 2000

0.91250000E 01 0.91750000E 01

C.WIGGLES FOR PANEL NO. 2 00000

0.97500000E 00 0.62500000E 00

000 PANEL NO. 3 INPUT VALUES 000

X1 = 2.000000 X2 = 10.000000 Y1 = 0.0 Z1 = 1.000000

Best Available Copy

K1 = 2.000000 K2 = 10.000000 Y1 = 1.000000 Z1 = 1.000000
 K3 = 10 K4 = 2 NUTEL = 1 NO. OF PANELS = 5

10 CUMULATIVE DIVISIONS FOR PANEL 3

0.0	0.12500000 00	0.25000000 00	0.37500000 00	0.50000000 00	0.62500000 00
0.75000000 00	0.87500000 00	0.93750000 00	0.98000000 01		

2 SPANWISE DIVISIONS FOR PANEL 3

0.0	0.10000000 01				
		20 *** XI ELEMENTS FOR PANEL 3***			
0.25000000 01	0.30000000 01	0.40000000 01	0.50000000 01	0.60000000 01	0.70000000 01
0.80000000 01	0.90000000 01	0.95000000 01	0.98000000 02	0.99000000 01	0.99500000 01
0.99900000 01	0.99990000 01	0.99995000 01	0.99998000 01	0.99999000 01	0.99999500 01
0.99999000 01	0.99999500 02				

		20 *** EFA ELEMENTS FOR PANEL 3***			
0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0
0.10000000 01	0.10000000 01	0.10000000 01	0.10000000 01	0.10000000 01	0.10000000 01
0.10000000 01	0.10000000 01				

20 ZFF ELEMENTS FOR PANEL NO. 3

0.10000000 01	0.10000000 01	0.10000000 01	0.10000000 01	0.10000000 01	0.10000000 01
0.10000000 01	0.10000000 01	0.10000000 01	0.10000000 01	0.10000000 01	0.10000000 01
0.10000000 01	0.10000000 01	0.10000000 01	0.10000000 01	0.10000000 01	0.10000000 01
0.10000000 01	0.10000000 01				

1 *** XI-J ELEMENTS FOR PANEL 3***

0.20000000 01

C.WIGGLES FOR PANEL NO. 3 *****

0.80000000 01

*** PANEL NO. 4 INPUT VALUES ***

K1 = 2.000000 K2 = 10.000000 Y1 = 1.000000 Z1 = 1.000000
 K3 = 2.000000 K4 = 10.000000 Y2 = 1.000000 Z2 = -1.000000
 K5 = 17 K6 = 3 NUTEL = 1 NO. OF PANELS = 5

10 CUMULATIVE DIVISIONS FOR PANEL 4

0.0	0.12500000 00	0.25000000 00	0.37500000 00	0.50000000 00	0.62500000 00
0.75000000 00	0.87500000 00	0.93750000 00	0.98000000 01		

1 SPANWISE DIVISIONS FOR PANEL 4

0.0	0.50000000 00	0.10000000 01			
		10 *** XI ELEMENTS FOR PANEL 4***			
0.20000000 01	0.30000000 01	0.40000000 01	0.50000000 01	0.60000000 01	0.70000000 01
0.80000000 01	0.90000000 01	0.95000000 01	0.98000000 02	0.99000000 01	0.99500000 01
0.99900000 01	0.99990000 01	0.99995000 01	0.99998000 01	0.99999000 01	0.99999500 01
0.99999000 01	0.99999500 02				

		10 *** EFA ELEMENTS FOR PANEL 4***			
0.10000000 01	0.10000000 01	0.10000000 01	0.10000000 01	0.10000000 01	0.10000000 01
0.10000000 01	0.10000000 01	0.10000000 01	0.10000000 01	0.10000000 01	0.10000000 01
0.10000000 01	0.10000000 01	0.10000000 01	0.10000000 01	0.10000000 01	0.10000000 01
0.10000000 01	0.10000000 01	0.10000000 01	0.10000000 01	0.10000000 01	0.10000000 01
0.10000000 01	0.10000000 01				

10 ZLE ELEMENTS FOR PANEL NO. 4

0.10000000 01	0.10000000 01	0.10000000 01	0.10000000 01	0.10000000 01	0.10000000 01
0.10000000 01	0.10000000 01	0.10000000 01	0.10000000 01	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	-0.10000000 01	-0.10000000 01	-0.10000000 01	-0.10000000 01
-0.10000000 01	-0.10000000 01	-0.10000000 01	-0.10000000 01	-0.10000000 01	-0.10000000 01

2 *** XI-J ELEMENTS FOR PANEL 4***

0.20000000 01

C.WIGGLES FOR PANEL NO. 4 *****

0.80000000 01

*** PANEL NO. 5 INPUT VALUES ***

40 = 10 45 = 2 40011 = 1 40112 PANELS = 5

10 CHRONOLOGICAL DIVISIONS FOR PANEL 9

0.14207081E 00 0.47400000E 00 0.43750000E 00 0.10700000E 01

2 SPANISH DIVISIONS FOR PANEL 5

0.100000000000000000

20 SET ELEMENTS FOR PANEL NO. 7

-0.1000000000 01 -0.1000000000 01 -0.1000000000 01 -0.1000000000 01 -0.1000000000 01 -0.1000000000 01

1 000 KI-J ELEMENTS FOR PANEL 5000

0.2000000000 01

C.WIGLES FOR PANEL NO. 5 00000

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12 15 25 41 52

5 DIEDRAL ANGLES FOR ALL PANELS

7.0	0.0	0.0	-0.8949944F (2)	0.18070000(0)
-----	-----	-----	-----------------	----------------

52 000 KI-V ELEMENTS ALL PANELS 000

0.44791641E 01	0.53458234E 01	0.63124990E 01	0.49374971E 01	0.56874933E 01	0.64175000E 0
----------------	----------------	----------------	----------------	----------------	---------------

92 000 XI-1 ELEMENTS ALL PANELS 000

0.42499971F 01 0.52499914E 01 0.62499970F 01 0.47083311F 01 0.55416594E 01 0.63790000F 01

02 000 XI-2 ELEMENTS ALL PANELS 000

12 2-V ELEMENTS FOR ALL PANELS					
0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.10000000E 01	0.10000000E 01
0.10000000E 01	0.10000000E 01	0.10000000E 01	0.10000000E 01	0.10000000E 01	0.10000000E 01

92 7-8 ELEMENTS WITH ALL PARTS

52 P-2 ELEMENTS WITH ALL PARTS

***** SPECIAL FILM, FOR ALL PANELS *****

52 NOT WIDTHS FOR ALL PAGES

T-S FOR THE 10 STOPS

ONLY-5 FOR THE 10 SERIES

2-3 FOR THE 10 STRIPS

```

0.0      0.0      0.0      0.0      0.0      0.0
0.10000000E 01  0.30000000E 02  -0.30000000E 00  -0.10000000E 01
                                DELT-S FOR THE 10 STRIPS

0.0      0.0      0.0      0.0      0.0      0.0
0.0      0.10000000E 01  0.10000000E 01  0.0
0.12500000E 00  0.17500000E 00  0.62500000E 00  0.87500000E 00  0.25000000E 00  0.75000000E 00
0.90000000E 00  0.25000000E 00  0.75000000E 00  0.90000000E 00

*** BODY NO. 1 INPUT VALUES ***
CENTER OF BODY COORDINATES X = 0.0      Z = 0.0
YPLAG = 0 ZPLAG = 1      SHAPE COEFFICIENT = 1.000000
BODY DIA LIMITS = 17 52

11 BODY ELEMENTS FOR BODY NO. 1
0.0      0.10000000E 01  0.20000000E 01  0.30000000E 01  0.40000000E 01  0.50000000E 01
0.60000000E 01  0.70000000E 01  0.80000000E 01  0.90000000E 01  0.10000000E 02

11 BODY LAULT FOR BODY NO. 1
0.0      0.69999999E 03  0.79999999E 00  0.89999999E 00  0.99999999E 00  0.10000000E 01
0.99999999E 00  0.79999999E 00  0.69999999E 00  0.50000000E 00  0.0

10 X ELEMENTS FOR ALL BODIES
0.50000000E 00  0.15000000E 01  0.25000000E 01  0.35000000E 01  0.45000000E 01  0.55000000E 01
0.65000000E 01  0.75000000E 01  0.85000000E 01  0.95000000E 01

10 Y ELEMENTS FOR ALL BODIES
0.50000000E 00  0.15000000E 01  0.25000000E 01  0.35000000E 01  0.45000000E 01  0.55000000E 01
0.65000000E 01  0.75000000E 01  0.85000000E 01  0.95000000E 01

10 Z ELEMENTS FOR ALL BODIES
0.0      0.0      0.0      0.0      0.0      0.0
0.0      0.0      0.0      0.0      0.0      0.0

10 W ELEMENTS FOR ALL BODIES
0.0      0.0      0.0      0.0      0.0      0.0
0.0      0.0      0.0      0.0      0.0      0.0

10 H ELEMENTS FOR ALL BODIES
0.10000000E 01  0.10000000E 01  0.10000000E 01  0.10000000E 01  0.10000000E 01  0.10000000E 01
0.10000000E 01  0.10000000E 01  0.10000000E 01  0.10000000E 01  0.10000000E 01  0.10000000E 01

10 M ELEMENTS FOR ALL BODIES
0.16000000E 00  0.76000000E 03  0.86000000E 00  0.92000000E 00  0.97000000E 00  0.99000000E 00
0.99000000E 00  0.76000000E 03  0.86000000E 00  0.92000000E 00  0.97000000E 00  0.99000000E 00

END OF PART1 - BASIC DATA CALCULATIONS

```

PARAM	NAME	Y	X	REQ
1	1	0	0	-5.0000
1	1	1	0	5.0000
2	1	0	0	-5.0000
2	1	1	0	5.0000

1	2	0	0	1.0000
2	2	0	0	1.0000

THE 52 B-MATRIX ELEMENTS FOR MODE NO. 1

-0.19297779E-01	0.14511675E-01	0.44124921E-01	-0.14753939E-02	0.20624917E-01	0.41124684E-01
0.02154626E-02	0.22467250E-01	0.16654198E-01	0.14235880E-01	0.21180727E-01	0.28124455E-01
0.24101448E-01	0.41757774E-01	0.35664763E-01	0.34570313E-01	-0.11000001E 00	-0.69499999E-01
-0.30000000E-01	0.49999999E-02	0.44979747E-01	0.84499999E-01	0.12999999E 00	0.82499999E-01
0.42499999E-01	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.11000001E 00	0.64499999E-01	0.80000000E-01	-0.49999999E-02	-0.49999999E-01
-0.49999999E-01	-0.12999999E 00	-0.82499999E-01	-0.84499999E-01		

THE 52 B-MATRIX ELEMENTS FOR MODE NO. 1

0.49000000E 01	-0.31255244E 00	0.50000000E 01	0.42707920E 01	0.50000000E 01	0.84541578E 01
0.50000000E 01	0.15624423E 01	0.50000000E 01	0.51124686E 01	0.50000000E 01	0.40624409E 01
0.50000000E 01	0.14174474E 01	0.50000000E 01	0.63541412E 01	0.50000000E 01	0.92704111E 01
0.50000000E 01	0.51124686E 01	0.50000000E 01	0.73454158E 01	0.50000000E 01	0.44741412E 01
0.50000000E 01	0.22244625E 02	0.50000000E 01	0.24453110E 02	0.50000000E 01	0.23044475E 02
0.50000000E 01	0.24449999E 02	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0

THE 52 B-MATRIX ELEMENTS FOR MODE NO. 2

0.36666778E-01	0.36666778E-01	0.36666778E-01	0.27499973E-01	0.36666778E-01	0.27499973E-01
0.23333373E-01	0.23333373E-01	0.23333373E-01	0.14666668E-01	0.14666668E-01	0.14666668E-01
0.17499999E-01	0.17499999E-01	0.17499999E-01	0.12499999E-01	0.12499999E-01	0.12499999E-01
0.14666668E-01	0.14666668E-01	0.14666668E-01	0.12499999E-01	0.12499999E-01	0.12499999E-01
0.20000000E-01	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0
0.0	-0.14666668E-01	-0.14666668E-01	-0.12499999E-01	-0.12499999E-01	-0.12499999E-01
-0.14666668E-01	-0.14666668E-01	-0.14666668E-01	-0.12499999E-01	-0.12499999E-01	-0.12499999E-01

THE 52 B-MATRIX ELEMENTS FOR MODE NO. 2

0.0	0.50000000E 01	0.0	0.50000000E 01	0.0	0.50000000E 01
0.0	0.50000000E 01	0.0	0.50000000E 01	0.0	0.50000000E 01
0.0	0.50000000E 01	0.0	0.50000000E 01	0.0	0.50000000E 01
0.0	0.50000000E 01	0.0	0.50000000E 01	0.0	0.50000000E 01
0.0	0.50000000E 01	0.0	0.50000000E 01	0.0	0.50000000E 01
0.0	0.50000000E 01	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0

U.W	U.W	U.W	U.W	U.W	U.W
0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0

BUOY HUMB N W ARJ

1	1	0	0	-5.0000
1	1	1	0	-5.0000
1	2	0	0	1.0000

THE 10 B-MATRIX ELEMENTS FOR MODE NO. 1

-0.67999999E-01	-0.10499999E-00	-0.44999999E-01	-0.55499999E-01	-0.19500000E-01	0.18999999E-01
0.50999999E-01	0.74999999E-01	0.83299999E-01	0.44999999E-01		

THE 10 WJ-MATRIX ELEMENTS FOR MODE NO. 1

0.50000000E-01	-0.22500000E-02	0.53700000E-01	-0.17500000E-02	0.50000000E-01	-0.12500000E-02
0.50000000E-01	-0.75000000E-01	0.50000000E-01	-0.25000000E-01	0.50000000E-01	0.74999999E-01
0.50000000E-01	0.74999999E-01	0.50000000E-01	0.12500000E-02	0.50000000E-01	0.17499999E-02
0.50000000E-01	0.22500000E-02				

THE 10 WJ-PRIME ELEMENTS FOR MODE NO. 1

0.0	0.50000000E-01	0.0	0.50000000E-01	0.0	0.50000000E-01
0.0	0.50000000E-01	0.0	0.50000000E-01	0.0	0.50000000E-01
0.0	0.50000000E-01	0.0	0.50000000E-01	0.0	0.50000000E-01
0.0	0.50000000E-01				

COL. NO. 1 UP DEL-CP MATRIX

0.46731171E-02	-0.97966523E-02	0.44374754E-02	0.12546356E-02	0.34520996E-02	0.12449531E-02
0.23169999E-02	0.74703506E-02	0.42244144E-01	0.27865108E-02	-0.19607878E-02	0.74774523E-02
-0.23169999E-02	0.21291150E-02	-0.12591474E-02	0.15737445E-02	-0.39249852E-02	-0.31615768E-01
-0.15179195E-02	-0.42811787E-02				

COLUMN NO. 1 OF THE 1004 MATRIX

0.25195145E-01	-0.50428762E-01	0.76276155E-01	-0.11514839E-01	0.17455246E-01	-0.88432014E-00
-0.76441551E-00	-0.26761483E-01	-0.11504444E-01	-0.23530371E-01	0.64128828E-00	-0.19685144E-01
-0.13914610E-01	-0.44865171E-00	-0.11610011E-01	-0.33448871E-00	-0.84853542E-00	-0.10662644E-01
-0.44865171E-00	0.13044339E-00	-0.13257143E-01	0.27196729E-00	-0.10912159E-01	0.52343429E-01
0.13044339E-01	0.13264852E-01	0.11741146E-01	0.23404674E-01	0.15545197E-01	0.49696207E-00
0.13264852E-01	0.54411064E-00	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0

THE 10 A-MATRIX ELEMENTS FOR MODE NO. 2

0.11999999E-01	0.77999999E-01	0.11999999E-01	0.34999999E-01	0.14999999E-01	0.17999999E-01
0.31999999E-01	0.77999999E-01	0.21999999E-01	0.49999999E-02		

THE 10 WJ-MATRIX ELEMENTS FOR MODE NO. 2

0.0	0.50000000E-01	0.0	0.50000000E-01	0.0	0.50000000E-01
0.0	0.50000000E-01	0.0	0.50000000E-01	0.0	0.50000000E-01
0.0	0.50000000E-01	0.0	0.50000000E-01	0.0	0.50000000E-01
0.0	0.50000000E-01				

THE 10 WJ-PRIME ELEMENTS FOR MODE NO. 2

0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0				

COL. NO. 2 UP DEL-CP MATRIX

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COLUMBIA 40. 2 to the AUGUST 1941

[illegible]

432 433 434

01	02	03	04	05	06	07	08	09	10	11	12	13	14	15	16	17	18	19	20	21	22	23	24	25	26	27	28	29	30	31	32	33	34	35	36	37	38	39	40	41	42	43	44	45	46	47	48	49	50	51	52	53	54	55	56	57	58	59	60	61	62	63	64	65	66	67	68	69	70	71	72	73	74	75	76	77	78	79	80	81	82	83	84	85	86	87	88	89	90	91	92	93	94	95	96	97	98	99	00	01	02	03	04	05	06	07	08	09	10	11	12	13	14	15	16	17	18	19	20	21	22	23	24	25	26	27	28	29	30	31	32	33	34	35	36	37	38	39	40	41	42	43	44	45	46	47	48	49	50	51	52	53	54	55	56	57	58	59	60	61	62	63	64	65	66	67	68	69	70	71	72	73	74	75	76	77	78	79	80	81	82	83	84	85	86	87	88	89	90	91	92	93	94	95	96	97	98	99	00	01	02	03	04	05	06	07	08	09	10	11	12	13	14	15	16	17	18	19	20	21	22	23	24	25	26	27	28	29	30	31	32	33	34	35	36	37	38	39	40	41	42	43	44	45	46	47	48	49	50	51	52	53	54	55	56	57	58	59	60	61	62	63	64	65	66	67	68	69	70	71	72	73	74	75	76	77	78	79	80	81	82	83	84	85	86	87	88	89	90	91	92	93	94	95	96	97	98	99	00	01	02	03	04	05	06	07	08	09	10	11	12	13	14	15	16	17	18	19	20	21	22	23	24	25	26	27	28	29	30	31	32	33	34	35	36	37	38	39	40	41	42	43	44	45	46	47	48	49	50	51	52	53	54	55	56	57	58	59	60	61	62	63	64	65	66	67	68	69	70	71	72	73	74	75	76	77	78	79	80	81	82	83	84	85	86	87	88	89	90	91	92	93	94	95	96	97	98	99	00	01	02	03	04	05	06	07	08	09	10	11	12	13	14	15	16	17	18	19	20	21	22	23	24	25	26	27	28	29	30	31	32	33	34	35	36	37	38	39	40	41	42	43	44	45	46	47	48	49	50	51	52	53	54	55	56	57	58	59	60	61	62	63	64	65	66	67	68	69	70	71	72	73	74	75	76	77	78	79	80	81	82	83	84	85	86	87	88	89	90	91	92	93	94	95	96	97	98	99	00	01	02	03	04	05	06	07	08	09	10	11	12	13	14	15	16	17	18	19	20	21	22	23	24	25	26	27	28	29	30	31	32	33	34	35	36	37	38	39	40	41	42	43	44	45	46	47	48	49	50	51	52	53	54	55	56	57	58	59	60	61	62	63	64	65	66	67	68	69	70	71	72	73	74	75	76	77	78	79	80	81	82	83	8
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13	0.93125	0.25000	1.00000	0.50000	0.91834800 01	0.10957275 02
14	0.85625	2.25000	1.00000	0.50000	0.10406773 01	0.20130270 01
15	0.78125	8.25000	1.00000	0.50000	0.29603653 01	-0.27007497 01
16	0.69063	9.12500	1.00000	0.50000	0.36076508 02	0.29843506 02
17	0.59113	9.42500	1.00000	0.50000	0.90457197 01	0.36407101 02
18	0.43125	2.25000	1.00000	-0.50000	-0.52471690 01	-0.13068812 01
19	0.35625	3.25000	1.00000	-0.50000	-0.14151204 00	0.26371018 00
20	0.28125	4.25000	1.00000	-0.50000	-0.40712671 01	0.39514776 01
21	0.19063	5.25000	1.00000	-0.50000	-0.13770022 02	-0.60473562 01
22	0.51125	6.25000	1.00000	-0.50000	-0.93837099 01	-0.19457306 02
23	0.43625	7.25000	1.00000	-0.50000	-0.10407286 01	-0.29151889 01
24	0.36125	8.25000	1.00000	-0.50000	-0.29407671 01	0.27047161 01
25	0.29063	9.12500	1.00000	-0.50000	-0.36076508 02	-0.29843513 02
26	0.19113	9.42500	1.00000	-0.50000	-0.90457194 01	-0.36408909 02
27	0.03125	2.25000	0.50000	-1.00000	-0.13515166 00	0.57133595 00
28	0.15625	3.25000	0.50000	-1.00000	-0.51851849 00	-0.89995731 00
29	0.28125	4.25000	0.50000	-1.00000	0.36425133 00	-0.13137350 01
30	0.40625	5.25000	0.50000	-1.00000	0.52444482 01	0.30017004 01
31	0.53125	6.25000	0.50000	-1.00000	-0.25449181 01	0.45314586 01
32	0.65625	7.25000	0.50000	-1.00000	-0.46352749 01	0.66281506 01
33	0.78125	8.25000	0.50000	-1.00000	-0.14527224 01	-0.46493271 01
34	0.85625	9.12500	0.50000	-1.00000	0.42646903 00	0.14574160 02
35	0.93113	9.42500	0.50000	-1.00000	-0.95442464 01	0.11491247 02

MODE M1 1				LIFT COEFFICIENT		MOMENT COEFFICIENT	
STEP NO.	X	Z	VIS				
1	1.5000	0.0	0.5000	11.070717	29.141011	-11.467719	-21.891312
2	2.5000	0.0	0.5000	10.319186	27.614761	-12.608169	-18.524979
3	3.5000	0.0	0.5000	27.449479	20.744045	-8.455248	-15.162904
4	4.5000	0.0	0.5000	11.999816	23.635219	-2.158457	-11.430099
5	1.5000	0.0	0.5000	67.118027	137.170753	-5.877938	-39.599060
6	2.5000	0.0	0.5000	87.183362	126.249636	-14.056466	-36.922958
7	0.5000	1.0000	0.1000	1.576412	-3.715916	-1.248104	2.718618
8	1.0000	0.5000	0.2000	7.248824	7.191329	-4.551122	-5.458472
9	1.0000	-0.5000	0.2000	-7.248796	-7.191346	4.551102	5.458501
10	0.5000	-1.0000	0.1000	-1.524467	3.715467	1.248072	-2.718609

MODE M1 1				LIFT COEFFICIENT		MOMENT COEFFICIENT	
MODE NO.	X	Z	VIS				
1	0.0	0.0	0.0	0.501440	1.761999	0.789920	-0.697967
CF =	65.901274	52.520111		CF =	0.000029	-0.000019	
CM =	-125.104490	-193.006531		CM =	0.0	0.0	
CL (ROLL) =	0.0	0.0					

PRESSURE MODE		INJECTION MODE		GENERALIZED FORCES	
1		1		0.44770924 01	0.39491491 02
1		2		7.14688185 02	0.14406627 02

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DATA NO.		TIME		PRESSURE	
1	2	3	4	5	6
1	0.00111	0.00111	1.00000	0.0	0.00000000 01
2	0.00111	0.00111	1.00000	0.0	-0.00000000 02
3	0.00111	0.00111	1.00000	0.0	-0.00000000 03
4	0.00111	0.00111	1.00000	0.0	0.00000000 04
5	0.00111	0.00111	1.00000	0.0	-0.00000000 05
6	0.00111	0.00111	1.00000	0.0	-0.00000000 06
7	0.00111	0.00111	1.00000	0.0	0.00000000 07
8	0.00111	0.00111	1.00000	0.0	0.00000000 08
9	0.00111	0.00111	1.00000	0.0	-0.00000000 09
10	0.00111	0.00111	1.00000	0.0	-0.00000000 10
11	0.00111	0.00111	1.00000	0.0	0.00000000 11
12	0.00111	0.00111	1.00000	0.0	-0.00000000 12
13	0.00111	0.00111	1.00000	0.0	-0.00000000 13
14	0.00111	0.00111	1.00000	0.0	0.00000000 14
15	0.00111	0.00111	1.00000	0.0	0.00000000 15
16	0.00111	0.00111	1.00000	0.0	-0.00000000 16
17	0.00111	0.00111	1.00000	0.0	-0.00000000 17
18	0.00111	0.00111	1.00000	0.0	0.00000000 18
19	0.00111	0.00111	1.00000	0.0	0.00000000 19
20	0.00111	0.00111	1.00000	0.0	-0.00000000 20
21	0.00111	0.00111	1.00000	0.0	-0.00000000 21
22	0.00111	0.00111	1.00000	0.0	0.00000000 22
23	0.00111	0.00111	1.00000	0.0	-0.00000000 23
24	0.00111	0.00111	1.00000	0.0	-0.00000000 24
25	0.00111	0.00111	1.00000	0.0	0.00000000 25
26	0.00111	0.00111	1.00000	0.0	0.00000000 26
27	0.00111	0.00111	1.00000	0.0	-0.00000000 27
28	0.00111	0.00111	1.00000	0.0	-0.00000000 28
29	0.00111	0.00111	1.00000	0.0	0.00000000 29
30	0.00111	0.00111	1.00000	0.0	0.00000000 30
31	0.00111	0.00111	1.00000	0.0	-0.00000000 31
32	0.00111	0.00111	1.00000	0.0	-0.00000000 32
33	0.00111	0.00111	1.00000	0.0	0.00000000 33
34	0.00111	0.00111	1.00000	0.0	0.00000000 34
35	0.00111	0.00111	1.00000	0.0	-0.00000000 35
36	0.00111	0.00111	1.00000	0.0	-0.00000000 36
37	0.00111	0.00111	1.00000	0.0	0.00000000 37
38	0.00111	0.00111	1.00000	0.0	0.00000000 38
39	0.00111	0.00111	1.00000	0.0	-0.00000000 39
40	0.00111	0.00111	1.00000	0.0	-0.00000000 40
41	0.00111	0.00111	1.00000	0.0	0.00000000 41
42	0.00111	0.00111	1.00000	0.0	0.00000000 42
43	0.00111	0.00111	1.00000	0.0	-0.00000000 43
44	0.00111	0.00111	1.00000	0.0	-0.00000000 44
45	0.00111	0.00111	1.00000	0.0	0.00000000 45
46	0.00111	0.00111	1.00000	0.0	0.00000000 46
47	0.00111	0.00111	1.00000	0.0	-0.00000000 47
48	0.00111	0.00111	1.00000	0.0	-0.00000000 48
49	0.00111	0.00111	1.00000	0.0	0.00000000 49
50	0.00111	0.00111	1.00000	0.0	0.00000000 50
51	0.00111	0.00111	1.00000	0.0	-0.00000000 51
52	0.00111	0.00111	1.00000	0.0	-0.00000000 52

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MODE NO. 2							
STRIP NO.	Y	Z	YYS	LIFT COEFFICIENT		MOMENT COEFFICIENT	
1	1.5000	0.0	0.0000	-12.532215	25.175320	8.659061	-10.182670
2	2.5000	0.0	0.0000	-8.600985	26.791851	5.662626	-11.602909
3	3.5000	0.0	0.0000	1.515256	26.006271	2.821109	-10.609795
4	4.5000	0.0	0.0000	2.626340	18.026368	1.525789	-6.900922
5	5.5000	0.0	0.0000	16.756325	52.986598	-1.789643	-14.150756
6	6.5000	0.0	0.0000	10.167522	51.065338	-5.610915	-13.999779
7	7.5000	1.0000	0.0000	2.021996	0.018154	-1.375269	-0.026756
8	8.5000	0.5000	0.2500	-0.155633	8.963908	-0.261762	-2.970565
9	9.5000	-0.5000	0.2500	0.155633	-8.963908	0.261765	2.970569
10	0.5000	-1.0000	0.0000	-2.021951	-0.018154	1.375261	0.026761

MODE NO. 2							
MODE NO.	Y	Z	YYS	LIFT COEFFICIENT		MOMENT COEFFICIENT	
1	0.0	0.0	0.0	-1.929580	-0.000000	0.806667	0.189516

CL 1	-5.264211	55.231779	CL 2	0.130069	0.000325
CM 1	10.027156	-111.647176	CM 2	0.0	0.0
CL 10001	0.0	0.0			

PRESSURE MODE	DEFLECTED MODE	GENERALIZED FORCES	
2	1	3.67429809E 01	0.15979133E 02
2	2	-0.26496241E 01	0.10273367E 02

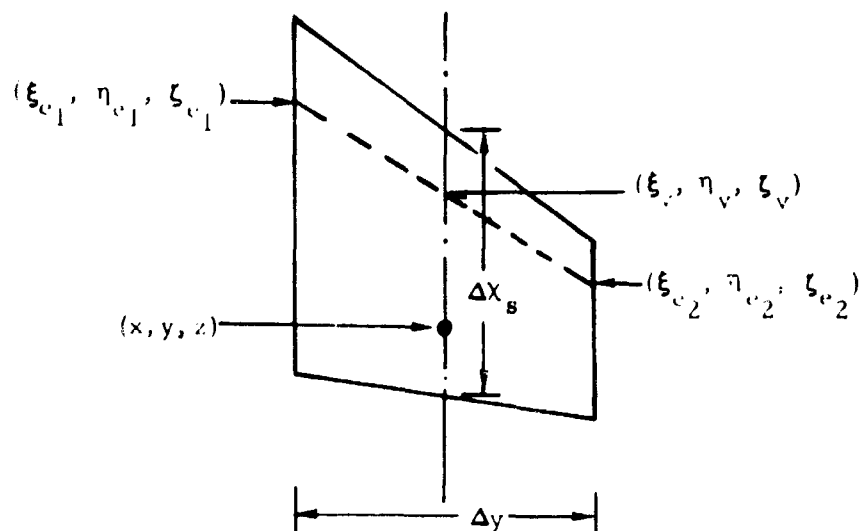
2.3 Blank and Labeled Common Blocks

2.3.1 Blank Common Block

Item No.	Mnemonics	Dimension	Symbol	Source	Description/Equation
1	NCNSM1, NBØX*			PART 1	Total number of boxes (points) on all lifting surfaces (panels) $\sum_{i=1}^N (nc_i - 1) (ns_i - 1),$ where $N = NØPAN =$ number of panels
2	NB			MAIN	Total number of bodies - input
3	NDELT		δ	MAIN	Symmetry flag - input
4	NDATA, N3			MAIN	Data flag - input
5	NØPAN, NP			MAIN	Total number of panels - input
6	IQ		r	initialized in MAIN, used in PART 2	Index of receiving point
7	IR		s		Index of sending point
8	JSPECS, JSP		ϵ	MAIN	Second symmetry flag - input
9	FMACH, M		M	MAIN	Mach Number - input
10	NCARAY	20		PART 1	Array of the number of the chordwise divisions for all panels

* A slash through the O indicates the letter O rather than the number zero.

Item No.	Mnemonics	Dimension	Symbol	Source	Description/Equation
11	NSARRAY	20		PART 1	Array of the number of the spanwise divisions for all panels
12	NBARAY	20	nba_i	PART 1	$nba_i = \sum_{j=1}^i (nc_j - 1) (ns_j - 1),$ where $i = 1, N\emptyset PAN$
13	ACAP, REFARA		A, S	MAIN	Reference area - input
14	B2, S		b/2, s	MAIN	Reference semispan-input
15	FL, REFCHD		\bar{c} , $2b_r$	MAIN	Reference chord - input
16	BETA		β	MAIN	$\beta = \sqrt{1 - M^2}$, where M = Mach Number
17	PI		π	MAIN	$\pi = 3.1415926$, built-in constant
18	KR, RFREQ		k_r	MAIN	Reduced frequency - input as RFREQ(20)
19	KRDBR		k_r/b_r	MAIN	Normalized k_r
20	GMA	50	γ_i	PART 1	Array of dihedral angles for all panels in radians; $i = 1, N\emptyset PAN$
21	X	400	x	PART 1	Array of x-coordinates of receiving points (3/4 chord point of box) - see sketch on next page.



Sketch 2.3-1

Item No.	Mnemonics	Dimension	Symbol	Source	Description/Equation
22	Y	400	y	PART 1	Array of y-coordinates of receiving points
23	ZZ	400	z	PART 1	Array of z-coordinates of receiving points
24	Z1	400	ξ_{e1}	PART 1	Array of x-coordinates
25	P1	400	η_{e1}	PART 1	Array of y-coordinates
26	ZZ1	400	ζ_{e1}	PART 1	Array of z-coordinates
					} of inboard sending points
27	Z2	400	ξ_{e2}	PART 1	Array of x-coordinates
28	P2	400	η_{e2}	PART 1	Array of y-coordinates
29	ZZ2	400	ζ_{e2}	PART 1	Array of z-coordinates
					} of outboard sending points

Item No.	Mnemonics	Dimension	Symbol	Source	Description/Equation
30	EV	400	ξ_v	PART 1	Array of x-coordinates
31	PV	400	η_v	PART 1	Array of y-coordinates
32	ZV	400	ζ_v	PART 1	Array of z-coordinates
					} of center-line sending points
33	SDELX	400	Δx_s	PART 1	Array of average chordlengths of all boxes.
34	DELY	400	Δy	PART 1	Array of box widths.
					See sketch above for items no. 21-34.

The Blank Common is used by the following subroutines of the program: MAIN, PART1, GENQ, AUGW and PART2.

2.3.2 Labeled Common Blocks

Common Name	Source	User Subroutines
/BØDY/	PART1	GENQ, AUGW, GENF
/MØDE/	GENQ	AUGW
/DLM/	KERNEL	INCRØ
/NTPS/	MAIN	GENB, GENW, GENQ, PART2, SSLEÇX, CXSS, AIC, AERØ, FINAL, GENF
/PIGW/	PART1	GENB
/XXZ/	PART1	FINAL, GENF
/XYZ/	PART1	GENQ, AERØ
/YZY/	GENB	AERØ, FINAL

The following is a detailed description of the contents of each labeled common block.

Common Name	Mnemonics	Dimension	Symbol	Source	Description/Equation
BODY	R0	100	R_{oj}	PART1	Body radius distribution
	R0P	100	R'_{oj}	PART1	x-derivative of body radius distribution
	NBEA	20	nbe_i	PART1	$nbe_i = \sum_{j=1}^i (nf_j - 1),$ where nf_j is the number of body endpoints for the j-th body; $i = 1, NB$

Common Name	Mnemonics	Symbol	Description
MODE	N8(80)	N8	Flag for spanwise variables
	NARQ(80)	NA	Number of ARQ values for all panels and modes
	LARQ(120, 2)	(N, M)	Array of subscripts of all ARQ's
	ARQ(120)	ARQ	Array of ARQ's
	H(400)	[h]	[h] matrix
	DH1(400)	[dh/dx]	[dh/dx] matrix
	DH2(400)	[d ² h/dx ²]	[d ² h/dx ²] matrix
			input for special cases only
			See Input Data, Sec 2.1.2

Common Name	Mnemonics	Symbol	Source	Description/Equation
DLM	K10	$K_1^{(s)}$	KERNEL	Planar part of
	K20	$K_2^{(s)}$		Nonplanar part of
	K1RT1	$R\bar{K}_1$		$R\bar{K}_1 (\Delta K_1) T_1$
	K1IT1	$Im\bar{K}_1$		$Im (\Delta K_1) T_1$
	K2RT2P	$R\bar{K}_2$		$R\bar{K}_2 (\Delta K_2) T_2^*$
	K2IT2P	$Im\bar{K}_2$		$Im (\Delta K_2) T_2^*$
	K10T1			$K_1^{(s)} T_1$
	K20T2P			$K_2^{(s)} T_2^*$
	E2	e^2	INCRØ	e semi-width of sending box

Common Name	Mnemonics	Dimension	Symbol	Source	Description/Equation
NTPS	NTP1			MAIN	Variable names for logical tape units used in program. For present assignment of units see Sec. 2.4 (Logical Tape Units)
	NTP2				
	NTP3				
	NTP4				
	NTP5				
	NTP6				
	NTP7				
	NTP8				
	NTP9				
	NTP10				
PIGW	CT1	20	\tilde{C}_1	PART1	Array of chord-lengths for inboard edge of all panels; $\tilde{C}_1 = X_2 - X_1$
	CT2	20	\tilde{C}_ϕ		Array of chord-lengths for out-board edge of all panels; $\tilde{C}_\phi = X_4 - X_3$
	TS	50	τ_{avg}		$\tau_{avg_i} = \frac{\tau_i + \tau_{i+1}}{2}$ <p>where τ_i is the i-th fractional spanwise division of panel; $i = 1, NSTRIP$</p>

Common Name	Mnemonics	Dimension	Symbol	Source	Description/Equation
XXZ	XØC	400	x/c	PART I	x-location of load points as a fraction of local chord
	XIJ	50	x _{le}		x-coordinate of leading edge for center of strips
XYZ	YS	50		PART I	y-coordinate of centerline of strip
	DELYS	50			Δy for strip
	ZS	50			z-coordinate of centerline of strip
	DELZS	50			Δz for strip
	FCAMMA	50			Dihedral angle for strip, in radians
	CWIG	50			Average chord length of strip
YXY	X1A*	50		GENB	(Alternative #1 AIC) Elastic axis (ALT. #2) Location of first control point.
	X3A*	50			Location of third control point.
	X5A*	50			Location of rotation axis for control surface
	X7A*	50			Location of rotation axis for tab
	X2A*	50			Location of second control point

* All coordinates given in fractions of local chord

2.4 LOGICAL TAPE UNITS

Tape Number		User Subroutines
Symbol	Present Assignment	
NTP1	1	MAIN, GENQ, PART2, SSLECX, SØLVIT, QUAS, FUTSØL
NTP2	13	MAIN, QUAS, FUTSØL
NTP3	3	MAIN, GENW, GENQ, SSLECX, CXSS, SØLVIT, QUAS, FUTSØL, AIC, FINAL, GENF
NTP4	4	MAIN, GENQ, CXSS, SØLVIT, QUAS, FUTSØL
NTP7	12	AUGW, GENF
NTP8	8	MAIN, GENQ, QUAS, FUTSØL
NTP9	9	MAIN, GENQ, SØLVIT, QUAS, FUTSØL, AIC, AERØ
NTP10	10	MAIN, GENB, GENW, GENQ, SSLECX, CXSS, AIC, FINAL, GENF
LTAPE	11	QUAS, FUTSØL
MTAPE	7	GENQ
NAT	NTP2	QUAS, FUTSØL
RHSTAP	4	QUAS, FUTSØL

Logical tape units 5 and 6 are used throughout the program as the standard READ and WRITE units respectively.

Logical tape units MTAPE, NTP9 and LTAPE can be specified as master tapes. MTAPE can be used as modal input tape for special cases; see 2.5.6.4 for pertinent information. NTP9 is used for saving the AIC supercolumns when this is desired — see 2.5.19.6. LTAPE contains the quasi-inverse matrix for future solutions of cases with new modes only — see 2.5.17.6.

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2.5 DESCRIPTION OF SUBROUTINES

Program H7WC is operational on the IBM 7094 and 360/65, and the GE 635 computers. The main part of the program controls all the twenty-one subroutines of the program: PART1, GENB, GENW, GENQ, AUGW, TKER, PART2, INCRØ, KERNEL, IDF1, IDF2, SNPDF, SSLECX, CXSS, SOLVIT, QUAS, FUTSØL, AIC, AERØ, FINAL and GENF. The following is a description of each of the subroutines in the order of their use within the program.

2.5.1 MAIN

Functional Description

The MAIN part of the program reads the header card for each data set, the Mach number, the reference area, the reference semi-span and the reference chord as well as the control flags that define the various types of cases. MAIN also reads the array of reduced frequencies, and in case of an AIC-type analysis, additional control data is read, as well as the gust data, if computation of the harmonic gust coefficients is also desired. In case of a modal-type analysis, the control data is read for the modes to be used. Depending on the type of the case, MAIN selects the appropriate subroutines in an overall frequency-loop, except for PART1 and GENB (latter in case of AIC's only), which are independent of the reduced frequency.

The following is a summary of all data items used in MAIN, grouped according to their source. Detailed description of each input data item is given in Sec. 2.1.2. Other data items are described in Sec. 2.3 (Contents of Common) and/or Sec. 2.4 (Logical Tape Units) as indicated below.

Data from Cards

FMACH, ACAP, FL, B2, NDELT, NP, NB, NRF, NCØRE, N1, N2, N3, N4, RFREQ(20), NSTRIP, NB1, NB2, NPR1, NPR2, NPR3, JØBNØ, JØBNØ2, NGUST, JSPECS, NPC, NSV, NBV, NYAW, GZRØ, XZRØ, VEL, WG, LIM(50, 3), NMD, NTA, N5, N6, N7, NTP1, NTPM, NMTP, NMTB.

See Sec. 2.1.2 for description.

Data from Common

NCNSM1, NCARAY(20), NSARAY(20), NBARAY(20), GMA(50),
X(400), Y(400), ZZ(400), Z1(400), P1(400), ZZ1(400), Z2(400), P2(400),
ZZ2(400), EV(400), PV(400), ZV(400), SDELX(400), DELY(400) -
from Blank Common.

YS(50), DELYS(50), ZS(50), DELZS(50), FGAMMA(50), CWIG(50) -
from Common /XYZ/.

X1A(50), X3A(50), X5A(50), X7A(50), X2A(50) - from Common
/YZY/.

XØC(400), CWIG(400) - from Common /XXZ/.

CT1(20), CT2(20), TS(50) - from Common /PIGW/.

R0(100), R0P(100), NBEA(20) - from Common /BØDY/.

K10, K20, K1RT1, K1IT1, K2RT2P, K2IT2P, K10 I1, K10 T2P,
E2 - from Common /DLM/.

See Sec. 2.3 for details.

Data from Tapes

MNEMONICS	SYMBOL	TAPE NAME	DESCRIPTION
AUGM(530)	$[A^{ww}]$	NTP1	Matrix of complex downwash factors, read one row at a time. Dimension of matrix is $NBØX \times$ $NBØX$.
WQ(130)	$[W_q]$	NTP3	Matrix of complex polynomial mode right-hand sides, read one row (of length NMD) at a time. Dimension of matrix is $NBØX \times$ NMD.
NTPØLD		LTAPE	Identification number of case for which the quasi-inverse of the matrix of downwash factors was saved on LTAPE in a previous run. For special cases only.

See Sec. 2.4 for present assignment of logical tape units.

Data to Common

NB, NDELT, NDATA, NØPAN, IQ, IR, JSPECS, FMACH, ACAP, B2, FL, BETA, PI, KR, KRDBR - to Blank Common.

NTP1, NTP2, NTP3, NTP4, NTP5, NTP6, NTP7, NTP8, NTP9, NTP10 - to Common/NTPS/.

See Sec. 2.3 for details.

Data to Tapes

MNEMONICS	SYMBOL	TAPE NAME	DESCRIPTION
NTP1		LTAPE	Identification number of current case, used to identify the quasi-inverse that is to be saved on LTAPE. For special cases only, when the input data N5 is set to 1.
AUGM(530)	$[A^{ww}; W_q]$	NTP4	Augmented downwash matrix: matrix of downwash factors augmented by the matrix of polynomial mode right hand sides written one row (of length NBOX + NMD) at a time. Dimension of matrix is NBOX x (NBOX + NMD).

2.5.2 SUBROUTINE PART1 (JCUM, CØEFP, CØEFB, YIN, ZIN)

Functional Description

Subroutine PART1 reads the geometry input of the idealized aircraft, the desired number of spanwise and chordwise divisions per panel, and the fractional location of the division lines. The panels are subdivided into boxes, and arrays of coordinates are generated for all boxes of all panels. In the case when body interference is also desired, PART1 reads the geometry data for all bodies and generates the necessary body-coordinate arrays. The computed arrays are made available for subsequent calculations through the Blank Common block placed in the

appropriate user subroutines. Additional items of computed geometry are transmitted to MAIN via the argument list of PART1, and to several other subroutines via the Labeled Common blocks /PIGW/, /XXZ/ and /XYZ/.

Data from Cards

XCAP(4), YCAP(2), ZCAP(2), NC, NS, CØEFP(20), TH(50), TAU(50), ZSC, YSC, NF, NZ, NY, CØEFB(20), F(50), RAD(50).

See Sec. 2.1.2 for details.

Data from Common

NB, NDEL, NDATA, NØPAN, IQ, IR, JSPECS, FMACH, ACAP, B2, FL, BETA, PI, KR, KRDBR - from Blank Common Block.

See Sec. 2.3 for details.

Data from Tapes

None

Data to Common

NCNSM1, NCARAY(20), NSARAY(20), NBARAY(20), GMA(50), X(400), Y(400), ZZ(400), Z1(400), P1(400), ZZ1(400), Z2(400), P2(400), ZZ2(400), EV(400), PV(400), ZV(400), SDELX(400), DELY(400) - to Blank Common block.

YS(50), DELYS(50), ZS(50), DELZS(50), FGAMMA(50), CWIG(50) - to Common /XYZ/.

XØC(400), XIJ(50) - to Common /XXZ/.

CT1(20), CT2(20), TS(50) - to Common /PIGW/.

R0(100), R0P(100), NBEA(20), BGMA(20) - to Common /BØDY/.

Data to Tapes

None

Data in Argument List

MNEMONICS	SYMBOL	DESCRIPTION/EQUATION
JCUM, NSTRIP		Number of strips on all panels
CØEFP(20)		Array of panel-coefficients – input
CØEFB(20)		Array of body-coefficients – input
YIN(20)		Array of y-coordinates of inboard edge of panels
ZIN(20)		Array of z-coordinates of inboard edge of panels

2.5.3 SUBROUTINE GENB (EV, SDELX, DELY, X, NSTRIP, BETA, IMX, NPC, NP, NSARAY, LIM)

Functional Description

Subroutine GENB reads the AIC-type mode flags for all panels of the lifting surfaces. It also reads the array of fractional chord lengths for the various control points and hinge line locations at the inboard edge and at the outboard edge of each panel. Subroutine GENB computes arrays of coordinates to be used in GENB and GENW and saves them on logical tape unit NTP10. Depending on the setting of the flag NPC (input data), subroutine GENB computes the integration matrix [B] according to Alternative 1 (NPC = 0) or Alternative 2 (NPC = 1). (See Eq. 2.5-12, Pt I, Vol I.) The [B] matrix is saved on logical tape unit NTP10 in a partitioned format, and is used in subroutine AIC for the computation of the AIC supercolumns* and harmonic gust coefficient columns, when latter is also desired.

Data from Cards

NØP(20), IS(20, 7), XHI (20, 7), XHØ (20, 7).

* A supercolumn of a matrix is a matrix-column whose elements are the columns of the original matrix taken in order; i. e., the second column is below the first, the third column is below the second, etc. A square matrix of order N becomes a supercolumn matrix of size $N^2 \times 1$.

See Sec. 2.1.2 for details.

Data from Common

NTP1, NTP2, NTP3, NTP4, NTP5, NTP6, NTP7, NTP8, NTP9,
NTP10 - from Common /NTPS/.

CT1(20), CT2(20), TS(50) - from Common /PIGW/.

Data from Tapes

None

Data to Common

X1A(50), X3A(50), X5A(50), X7A(50), X2A(50) - to Common /YZY/.

Data to Tapes

MNEMONICS	SYMBOL	TAPE NAME	DESCRIPTION
NSTRIP			Total number of strips on all lifting surfaces (panels)- input data.
NSIZ B(j, 1)			Number of AIC-modes for strip j
NSIZ B(j, 2)			Number of boxes in strip j
NSIZ B(50, 2)			
LIM(50, 3)			Aerodynamic box limits - input
XISLCT(50, 10)		NTP10	AIC-type modal flags - input
XH(50, 7)			Array of AIC-type reference coordinates
X(400)	x		} See Blank Common
SDELX(400)	Δx		
B(30, 10)	[B]		Integration matrix for the Aerodynamic Influence Coefficients (AIC's).

Data in Argument List

MNEMONICS	SYMBOL	DESCRIPTION
EV(400)	η_v	} See Blank Common
SDELX(400)	Δx_s	
DELY(400)	Δy	
X(400)	x	
NSTRIP		Total number of strips - input
BETA	β	See Blank Common
IMX		Maximum number of rigid body modes for case
NPC		Camber flag - input
NP, NØPAN		Number of panels - input
NSARRAY(20)		Array of the number of spanwise divisions (NS) for all panels
LIM(50, 3)		Aerodynamic box limits for all strips - input

2. 5. 4 SUBROUTINE GENW (KRDBR, NCSFTØ, NPC, NGUST, WG, GZRØ, XZRØ)

Functional Description

Subroutine GENW computes the AIC-type substantial derivative matrix, [W] described in Sec. 2. 5. 1*. Depending on the setting of the flag NPC (input data), the [W] matrix is computed according to Alternative 1 (NPC = 0) or Alternative 2 (NPC = 1). The matrix is saved for subsequent calculations on logical tape unit NTP3 in a partitioned format, described in Sec. 2. 5. 2*. In the case when computation of the harmonic gust coefficients is also desired, subroutine GENW also computes the gust downwash vector, and saves it on logical tape unit NTP3.

*See Part I, Vol I

Subroutines GENB and GENW are used only if an AIC-type analysis is desired.

Data from Cards

None

Data from Common

YS(50), DELYS(50), ZS(50), DELZS(50), FGAMMA(50), CWIG(50) – from Common /XYZ/ .

Contents of Common /NTPS/.

Data from Tapes

NSTRIP, NSIZB(50, 2), LIM(50, 3), XISLCT(50, 10), XH(50, 7), X(400), SDELX(400) – from NTP10, written in GENB.

Data to Common

None

Data to Tapes

MNEMONICS	SYMBOL	TAPE NAME	DESCRIPTION
GRHS(400)	$\begin{Bmatrix} W \\ w_{gust} \end{Bmatrix}$	NTP3	<p>Gust right hand sides (downwash vector) $W_{gust,i} = \cos(\Gamma_g - \gamma_i) \cdot \exp(-i \frac{k_r}{b_r} (x_i - x_o))$, where</p> <p>$\Gamma_g$ = gust field declination from the vertical – input</p> <p>γ_i = dihedral angle of lifting surface at point i</p> <p>k_r = reduced frequency – input</p> <p>b_r = reference semichord</p> <p>x_i = x-coordinate of downwash point i</p> <p>x_o = gust reference coordinate – input</p>

MNEMONICS	SYMBOL	TAPE NAME	DESCRIPTION
W(30,10)	$[W_{AIC}]$	NTP3	Partitioned AIC-type downwash matrix. See Sec. 2.5.2*.

Data in Argument List

MNEMONICS	SYMBOL	DESCRIPTION
KRDBR		k_r/b_r
NCSFTØ, NBØX		Total number of boxes on all panels
NPC		Camber flag
NGUST		Gust flag
WG	W_g	Harmonic gust amplitude
GZRØ	Γ_g	Gust angle of declination
XZRØ	x_o	Gust reference coordinate

} input

*See Part I, Vol I

2.5.5

SUBROUTINE GENQ (N4, N5, N6, NRF, JRF, NBE, NMD,
NMTF, NMTB, NCØRE, KDØ2, YIN,
ZIN, CØEFP, CØEFB)

Functional Description

Subroutine GENQ generates the normalwash boundary condition for all lifting surface elements. It also generates the integration matrix used to obtain the generalized forces. The normalwash w_{ri} (r ranges over all lifting surface box receiving points while i ranges over all modes) is written on tape row by row ($r = 1, i = 1, 2, \dots, NMD, r = 2, i = 1, 2, \dots, NMD, \text{etc.}$) to facilitate its use with subroutine SØLVIT. However, if the number of modes times the number of unknowns (NCØRE) exceeds the size of the working array in SØLVIT, and/or if the input flag N5 is set to 1, the W matrix is written on tape in column order, preceded by the number of modes (NMD), ready for use by subroutines QUAS and FUTSØJL. The integration matrix B_{ri} is written on tape in column order.

The formula for w used is given by equations 2.3-2* and 2.3-3* for cases without bodies, and by equation 2.2-13* for body cases; the contribution of the body effect to the normalwash, w_{B_r} , is computed in subroutine AUGW, which is called from subroutine GENQ for body cases only. w_{B_r} is computed one column at a time, and is transmitted to GENQ via the argument list of subroutine AUGW. The total normalwash matrix for body cases, $(w_r - w_{B_r})_i$, is then written on tape in either row, or column order depending on the size (NCØRE) of the problem and/or on the setting of the input flag N5.

Data from Cards

NR, NQ, NARQ(80), N8(80), LARQ(120, 2) and ARQ(120) for standard polynomial cases (N6 = 0).

When modes are input from cards (N6 = 2), the data input from cards are:

NBØX, BQ(400), H(400) and EH1(400).

*See Part I, Vol I

Data from Common

Contents of Blank Common and the labeled common blocks
/NTPS/ and /BODY/.

Data from Tapes

N6 = 0 cases:

None, if NRF = 1. If NRF > 1 and JRF > 1 (see Sec. 2.5.5.), the following data items are read from tape NTP10:

BQ(400), N8(80), NARQ(80), LARQ(120, 2), and ARQ(120).

N6 = 1 cases: modes are input from tape as follows.

NBØX, BQ(400), H(400) and DH1(400) from tape MTAPE; present assignment is NTP8 = 8.

Data to Common

N8(80), NARQ(80), LARQ(120, 2), and ARQ(120),
H(400), DH1(400) and DH2(400) to common /MODE/.

Data to Tapes

MNEMONICS	SYMBOL	TAPE	DESCRIPTION
BQ(400)	$[B_q]$	NTP10	Integration matrix for modal analysis written in column order. Dimension of matrix is NTØT x NMD, where NTØT is the total number of elements (e.g., NTØT = NBØX, if there are no body elements, and NTØT = NBØX + NBE otherwise, where NBE = total number of body elements).
NMD	NM	NTP4	Number of modes; written only if N6 = 1 and/or NCØRE ≥ 4000.

MNEMONICS	SYMBOL	TAPE	DESCRIPTION
WQ(400)	$[W_q]$	NTPW	Matrix of polynomial mode right hand sides (normalwash). Dimension of matrix is $NB\emptyset X \times NMD$. The matrix $[W_q]$ is written on tape in row order, unless $N5 = 1$, and/or $NC\emptyset RE \geq 4000$; in latter case, $[W_q]$ is written on tape in column order. Note that NTPW = NTP3 in case of row order, and NTPW = NTP4 in case of column order.
BQ(400)	$[B_q]$	NTPB	The following items are written on tape only if more than one reduced frequency is prescribed for the case ($NRF > 1$).
N8(80)			See Input Data, 2.1.2
NARQ(80)	NA		
LARQ(120, 2)	(N, M)		
ARQ(120)	ARQ		Note that NTPB = NTP2 for cases without bodies, and NTPB = 10 for body cases.

Data in Argument List

MNEMONICS	SYMBOL	DESCRIPTION
N4		Print flag - input
N5		Flag to save quasi-inverse - input
N6		Modal input data flag - input
NRF		Number of reduced frequencies for case - input
JRF		Sequence number of reduced frequency currently used in the frequency do-loop

MNEMONICS	SYMBOL	DESCRIPTION
NBE		Total number of body elements
NMD	NM	Number of modes - input
NMTP	MP	Total number of ARQ - values for all panels and modes - input
NMTB	MB	Total number of ARQ - values for all bodies and modes - input
NCØRE		Total number of unknowns for case - input
KDØ2	KD/2	4000 - built in constant (MAIN)
YIN(20)	Y1	Array of y-coordinates of inboard edge of panels
ZIN(20)	Z1	Array of z-coordinates of inboard edge of panels.
CØEFP(20)	} coef	Array of modal coefficients for panels
CØEFB(20)		Array of modal coefficients for bodies

} See
Eq.(2. 3-3)

2.5.6 SUBROUTINE AUGW (NMD, NBOX, NBE, NMTB, N4, N6, NRF, JRF, J, CØEFB, ASUM)

Functional Description

This subroutine gives the incremental normalwash, w_{Br} , at all lifting surface boxes, r , due to the axial singularities on all bodies, B . Slender body theory is used to obtain the strength of the axial singularities. The formulae used in this subroutine are given by Equations(2.2-8)through (2.2-10)* The incremental normalwash values, w_{Br} , are transmitted to subroutine GENQ via the argument list under the mnemonic name ASUM one column (for one mode) at a time. This subroutine also computes the interaction matrix of all body elements for the calculation of the generalized forces for body cases.

Subroutine AUGW is bypassed for data without body elements.

*See Part I, Vol I

Data from Cards

None for the special case when $N6 = 1$ (data input from tape MTAPE).

For standard polynomial cases ($N6 = 0$) data input from cards is same as for subroutine GENQ, except that data is now read for body elements instead of panel boxes.

Data from Common

Contents of Blank Common, and the labeled common blocks /BØDY/ and /MØDE/.

Data from Tapes

None for standard polynomial mode cases. For the special case when the modal data flag $N6 = 1$, the following data items are input from tape:

MNEMONICS	SYMBOL	TAPE	DESCRIPTION
NBE			Number of body elements
BB(80)	$[B_q]$	NTP8	Integration matrix for the body elements
H(400)	$[h]$		$[h]$ matrix
DH1 (400)	$[dh/dx]$		$[dh/dx]$ matrix
DH2(400)	$[d^2h/dx^2]$		$[d^2h/dx^2]$ matrix
			The above three matrices are read in column order. Dimension of matrices is NBE x NMD.

Data to Common

None

Data to Tapes

MNEMONICS	SYMBOL	TAPE	DESCRIPTION
BB(80)	B_q	NTP8	One column of the integration matrix for body elements written on tape for one mode; length of column is NBE. The following items are written on tape only if more than one reduced frequency is prescribed for case (NRF > 1)
N8(80)	N8	NTPB	See Input Data 2.1.2
NARQ(80)	NA		
LARQ(120, 2)	(N, M)		
ARQ(120, 2)	ARQ		

Data in Argument List

MNEMONICS	SYMBOL	DESCRIPTION
NMD	NM	Number of modes - input
NBØX		Total number of boxes on all lifting surfaces
NBE		Total number of body elements
NMTB	MB	Total number of ARQ values for all bodies and modes - input

MNEMONICS	SYMBOL	DESCRIPTION
N4		Print flag for writing the [AP ^z], [AP ^y] matrices
N6		Modal input flag
NRF		Number of reduced frequencies for case
JRF		Sequence number of current frequency in the frequency do-loop
J		Sequence number of current mode in modal do-loop
CØEFB(20)	coef	Array of scale factors for body deflection modes.
ASUM(400)		Incremental [W] due to body interference

} input

2.5.7

SUBROUTINE TKER (X0, Y0, Z0, KR, BR, GAMS,
GAMSIG, KKR, KKI, FMACH)

Functional Description

Subroutine TKER evaluates the total unsteady kernel for one receiving-point/sending-point combination for body cases only. (See Eq. A. 5)* Subroutine TKER is essentially the same as subroutine KERNEL, except that the total, rather than the incremental oscillatory kernel is computed here, and that the planar and nonplanar parts of the kernel are not separated. Subroutine TKER is called from subroutine AUGW; it is bypassed in the case when the input data contains no bodies.

Subroutine TKER uses no card-, tape-, or common-data input or output; data transmission to and from TKER is done through its argument list.

*See Part I, Vol I

Data in Argument List

MNEMONICS	SYMBOL	DESCRIPTION
X0	x_o	$x_i - \xi_j$
Y0	y_o	$y_i - \eta_j$
Z0	z_o	$z_i - \xi_j$
KR	k_r	Reduced frequency
BR	b_r	Reference semi-chord
GAMS	γ_s	Dihedral angle of sending point j
GAMSIG	γ_r	Dihedral angle of receiving point i
FMACH, M	M	Mach Number
E2		e^2 , where e is semi-width of sending box
KKR		Real part of total (steady + unsteady) kernel
KKI		Imaginary part of total kernel

2.5.8 SUBROUTINE PART2 (NYAW, NBV)

Functional Description

Subroutine PART2 is used for the calculation of the matrix of downwash factors, [D]. (See Eq. 2.1-14.*) It prepares the arguments to subroutines SNPFD and INCRØ, which do the actual computation of the downwash factor elements, and calls these in a do-loop for all boxes of all lifting surfaces. The steady part of the downwash factor elements is computed in subroutine SNPFD, and the incremental oscillatory part in subroutine INCRØ; the call to INCRØ is bypassed for steady cases ($k_r = 0.0$). Subroutine PART 2 also accounts for the effect of the (two) symmetry plane(s).

Data from Cards

None

*See Part I, Vol I

Data from Common

Contents of Blank Common, Common /NTPS/, and Common /DLM/.

Data from Tapes

None

Data to Common

None

Data to Tapes

MNEMONICS	SYMBOL	TAPE	DESCRIPTION
AWW(400) AWWI(400)	$\left\{ \begin{array}{l} [A^{ww}] \\ [D] \end{array} \right.$	NTP1	Matrix of downwash factors written on tape in row order. Dimension of matrix is $NB\emptyset X \times NB\emptyset X$. The real parts of the matrix elements are called AWW in the program, and the imaginary parts are called AWWI.

Data in Argument List

NYAW - yaw flag
NBV - number of boxes on vertical panels
 (fin) of T-tail case: } input in MAIN

2.5.9 SUBROUTINE INCR \emptyset (AX, AY, AZ, AX1, AY1, AZ1, AX2, AY2, AZ2, GAMS, GAMSIG, LHS, IR, I \emptyset , NBXS, NCPNB, NDBLE, NBV, DELR, DELT, FL, BETA, SDELX, DELY, KR)

Functional Description

Subroutine INCR \emptyset prepares the argument for the subroutines KERNEL, IDF1 and IDF2. It calls subroutine KERNEL to compute the

incremental oscillatory part of the kernel, K , for each receiving-sending box combination at the three points of the bound vortex segment: at the center (K_c), at the inboard point (K_i) and at the outboard point (K_o). Since even a relatively small case requires many kernel computations (e. g., a wing with 100 boxes, in symmetry, requires $100 \times 300 \times 2 = 60,000$ kernel values), extra programming effort was made to save on computing time. Neighboring strips within a wing panel have a common kernel on the common boundary; this fact is utilized in INCRØ so that the computing time of the kernels is reduced by about 30 percent.

After the triplet of kernels, (K_c , K_i , K_o), is obtained for one receiving-sending box combination, subroutine INCRØ computes the coefficients of the parabolas for the numerical integrations done in two parts. The planar part of the integration is done in subroutine IDF1, and the nonplanar part in IDF2.

Subroutine INCRØ uses no card-, or tape input or output; transmission of data is done through its argument list and the labeled common /DLM/.

Data in Argument List

MNEMONICS	SYMBOL	DESCRIPTION/EQUATION
AX	x_o	$x - \xi_v$
AY	y_o	$y - \eta_v$
AZ	z_o	$z - \zeta_v$
AX1		$x - \xi_{e1}$
AY1		$y - \eta_{e1}$
AZ1		$z - \zeta_{e1}$
AX2		$x - \xi_{e2}$
AY2		$y - \eta_{e2}$
AZ2		$z - \zeta_{e2}$

See Sec. 2.3.1
(Blank Common)

MNEMONICS	SYMBOL	DESCRIPTION/EQUATION
GAMS	γ_r	Dihedral angle of receiving point i
GAMSIG	γ_s	Dihedral angle of sending point j
LHS		Flag; 0 when effect of right wing is computed, 1 for left-wing
IR	j	Index of sending point
IØ		An index running from 1 through NCMI, where NCMI is the number of chordwise boxes of the strip in which the sending point lies.
NBXS		The number of boxes on the wing panel in which the sending point lies + the total number of boxes of the preceding panels.
NCPNB		The number of boxes in the first strip of the panel in which the sending point lies + the total number of boxes of the preceding panels.
NDBLE, JSPECS, JSP	ϵ	Second symmetry flag - input
NBV		Total number of boxes on vertical panels of T-tail cases - input
DELR		Real part of the incremental oscillatory downwash factor
DELI		Imaginary part of the incremental oscillatory downwash factor
FL, REFCHD	\bar{c}	Reference chord - input
BETA	β	$\sqrt{1 - M^2}$
SDELX	Δx_s	Average box-chord of sending box
DELY		2c, box width of sending point
KR	k_r	Reduced frequency

2.5.10

SUBROUTINE KERNEL (X0, Y0, Z0, KR, BR, GAMS, GAMSIG,
EPS, T1, T2, M)

Functional Description

Subroutine KERNEL evaluates the modified incremental oscillatory kernel, K , for one receiving-sending point combination at a time. The planar and nonplanar parts of the kernel are evaluated separately and are returned to the calling program, subroutine INCRØ, via the labeled common /DLM/. (Evaluates right hand side of Eq. (2.1-13)*)

Subroutine KERNEL uses no card or tape input or output; transmission of data is done through the argument list and the labeled common /DLM/.

Data in Argument List

MNEMONICS	SYMBOL	DESCRIPTION/EQUATION
X0	x_o	AX for center AX1 for inboard edge AX2 for outboard edge
Y0	y_o	AY for center AY1 for inboard edge AY2 for outboard edge
Z0	z_o	AZ for center AZ1 for inboard edge AZ2 for outboard edge
KR	k_r	Reduced frequency
BR	$b_r, c/2$	Reference semichord
GAMS	y_r	} of sending box; see 2.5.9
GAMSIG	y_s	

*See Part I, Vol I

MNEMONICS	SYMBOL	DESCRIPTION/EQUATION
EPS	ϵ	$\epsilon = 0.00001$; built in constant from INCRØ
T1	T_1	$T_1 = \cos(\gamma_r - \gamma_s)$
T2	T_2	$T_2 = [z_o \cos \gamma_r - y_o \sin \gamma_r](z_o \cos \gamma_s - y_o \sin \gamma_s)/r^2$
M, FMACH	M	Mach number

2.5.11 SUBROUTINE IDF1(EF, E2, AT2, ETA01, ZET01, ARE,
AIM, BRE, BIM, CRE, CIM, RISQX,
XIJR, XIJI)

Functional Description

Subroutine IDF1 performs the integration of the planar parts of the incremental oscillatory kernels; see Eq. B. 9.* The result of the integration is the complex number (XIJR, XIJI), returned to subroutine INCRØ via the argument list of IDF1.

Subroutine IDF1 uses no card-, tape- or common input or output; data transmission to and from IDF1 is done through its argument list.

Data in Argument List

MNEMONICS	SYMBOL	DESCRIPTION
EE	$e, \Delta y_s/2$	Semiwidth of sending box j
E2	e^2	
AT2	$ T_2 $	See 2.5.10
ETA01	\bar{y}	$y_o \cos \gamma_r + z_o \sin \gamma_r$
ZET01	\bar{z}	$z_o \cos \gamma_r - y_o \sin \gamma_r$

*See Part I, Vol I

MNEMONICS	SYMBOL	DESCRIPTION
ARE	$\text{Re}(A_1)$	$\left\{ \begin{array}{l} \text{Coefficients of the parabola} \\ \text{for planar part of} \\ \text{kernel-integration} \end{array} \right.$
AIM	$\text{Im}(A_1)$	
BRE	$\text{Re}(B_1)$	
BIM	$\text{Im}(B_1)$	
CRE	$\text{Re}(C_1)$	
CIM	$\text{Im}(C_1)$	
RISQX	r^2	$\bar{y}^2 + \bar{z}^2$
XIJR		Real part of integral
XIJI		Imaginary part of integral
		$\left. \begin{array}{l} \text{Real part of integral} \\ \text{Imaginary part of integral} \end{array} \right\} \text{planar contribution}$

2.5.12 SUBROUTINE IDF2 (EE, E2, AT2, ETA01, ZET01, A2R, A2I, B2R, B2I, C2R, C2I, RISQX, DIJR, DIJI)

Functional Description

Subroutine IDF2 performs the integration of the nonplanar parts of the incremental oscillatory kernel; see Eqs. (B.15) and (B.16).^{*} The result of the integration is the complex number (DIJR, DIJI), returned to subroutine INCRØ through the argument list of IDF2.

Subroutine IDF2 uses no card, tape or common input or output; data transmission to and from IDF2 is done through its argument list.

^{*}See Part I, Vol I

Data in Argument List

MNEMONICS	SYMBOL	DESCRIPTION
EE	e	See 2.5.11
E2	e^2	
AT2	$ T_2 $	
ETA01	\bar{y}	Coefficients of the parabola for nonplanar part of kernel integration
ZET01	\bar{z}	
A2R	$\text{Re}(A_2)$	
A2I	$\text{Im}(A_2)$	See 2.5.11
B2R	$\text{Re}(B_2)$	
B2I	$\text{Im}(B_2)$	
C2R	$\text{Re}(C_2)$	Real part of integral
C2I	$\text{Im}(C_2)$	
R1SQX		
DIIJR		Imaginary part of integral
DIIJI		

2.5.13 SUBROUTINE SNPDEF (SL, CL, TL, SGS, CGS, SGR, CGR,
X0, Y0, Z0, EE, DIJ, BETA, CV)

Functional Description

Subroutine SNPDEF computes the steady downwash factors for one receiving-sending box combination at a time. The result, DIJ, is returned to PART2 via its argument list. (See Eq. C.35b, Part I, Vol I.)

Subroutine SNPDEF uses no card, tape, or common data input or output; data transmission to and from SNPDEF is done through its argument list.

Data in Argument List

MNEMONICS	SYMBOL	DESCRIPTION
SL		$\sin \Lambda$, where Λ is the sweep angle of the 1/4-chord line of sending box [sweep angle of bound vortex] modified by Prandtl-Glauert rule.
CL		$\cos \Lambda$
TL		$\tan \Lambda$
SGS		$\sin \gamma_s$
CGS		$\cos \gamma_s$
SGR		$\sin \gamma_r$
CGR		$\cos \gamma_r$
X0		} See 2.5.10
Y0		
Z0		
EE	e	Semi-width of sending box
DIJ		Steady contribution to downwash factor
BETA	β	$\sqrt{1 - M^2}$
CV	$c_{avg}, \Delta x_s$	Δx , average chord length of sending box

2.5.14

SUBROUTINE SSLECX (NCNSM1, MD, NGUST)Functional Description

Subroutine SSLECX is used with AIC-type data, and only if $NCORE < 4000$. It reads the matrix of downwash factors $[D]$ from tape NTP1 one row at a time, and augments it with the AIC-type downwash matrix $[W]_{AIC}$ read from tape NTP3. For gust cases, when $NGUST \neq 0$, subroutine SSLECX also reads the gust downwash vector from tape NTP3. The augmented matrix $[D|W]$ is saved on tape NTP4 in row order, ready to be used by subroutine SOLVIT to obtain the solutions to the system of simultaneous linear equations represented by the matrix $[D|W]$.

Data from Cards

None

Data from Common

Contents of Common /NTPS/.

Data from Tapes

MNEMONICS	SYMBOL	TAPE	DESCRIPTION
NSTRIP	[A]	NTP10	} See 2.5.3
NSIZ B(50, 2)			
LIM(50, 3)			
AUGM(530)			
GUST(400)	$\{W_{gust}\}$	NTP3	Gust downwash vector
W(30, 10)	$[W]_{AIC}$	NTP3	AIC type downwash (normalwash). } See 2.5.4

Data to Common

None

Data to Tapes

MNEMONICS	SYMBOL	TAPE	DESCRIPTION
AUGM(530)	[A W]	NTP4	Augmented downwash factor matrix

Data in Argument List

MNEMONICS	SYMBOL	DESCRIPTION
NCNSM1	n	Total number of boxes on all lifting surfaces (panels)
MD	m	Total number of AIC-type right-hand sides NSTRIP $m = \sum_{j=1}^{NSTRIP} XISLCT_{jmax}$ where NSTRIP = total number of strips on all lifting surfaces, and $XISLCT_{jmax}$ = the number of elastic modes for j-th strip - whenever NGUST = 0. When NGUST ≠ 0 $m = \sum_{j=1}^{NSTRIP} XISLCT_{jmax} + 1.$
NGUST		Gust flag - input

2.5.15

SUBROUTINE CXSS (NCNSM1, MD, NGUST)

Functional Description

Subroutine CXSS is used with AIC-type data, and only if $NCORE \geq 4000$, and/or $N5 = 1$. It reads the AIC-type downwash matrix, $[W]_{AIC}$, from tape NTP3 and forms columns of right-hand sides. When $NGUST \neq 0$, subroutine CXSS also reads the gust downwash vector from tape NTP3; this becomes the last column of right hand sides. These are saved on tape NTP4 preceded by the number of right hand sides for the case, ready to be used by subroutine QUAS and FUTSOL to obtain the solutions.

Data from Cards

None

Data from Common

Contents of Common /NTPS/.

Data from Tapes

MNEMONICS	SYMBOL	TAPE	DESCRIPTION
NSTRIP			} See 2.5.3
NSIZB(50, 2)		NTP10	
LIM(50, 3)			
GUST(400)	$\{W_g\}$	NTP3	Gust downwash vector
W(30, 10)	$[W]_{AIC}$	NTP3	Partitioned matrix of AIC-type downwash (normalwash) } see 2.5.4

Data to Common

None

Data to Tapes

MNEMONICS	SYMBOL	TAPE	DESCRIPTION
NMD	m	} NTP4	Number of AIC-type right hand sides (elastic modes), including the gust right hand side whenever NGUST \neq 0
WX(400)			Matrix of AIC-type right hand sides written in column order. Dimensions of matrix is NBØX x NMD

Data in Argument List

Same as in 2. 5. 14

2. 5. 16 SUBROUTINE SØLVIT (ND, MD, KD, NI, NM, NØ, NW, NPR1)

Functional Description

Subroutine SØLVIT is used for cases with NCØRE < 4000, and/or N5 = 0. It solves the system of simultaneous linear equations represented by the augmented matrix $[D | W]$, written on tape NTP4. The solutions are saved on tape NTP3 in column order; the data on the input tape, NTP4, is not preserved.

Data from Cards

None

Data from Common

None

Data from Tapes

AUGM(530), the augmented downwash factor matrix from tape NTP4 - see 2. 5. 1 or 2. 5. 14.

Data to Common

None

Data to Tapes

MNEMONICS	SYMBOL	TAPE	DESCRIPTION
A(4000)		NTP3	Solution matrix written in column order; dimension of matrix is $ND \times MD$, where $ND = NB\emptyset X$, and $MD = NMD$.

Note that subroutine $S\emptyset LVIT$ uses a single complex work array, A(4000), for all tape input and output variables.

Data in Argument List

MNEMONICS	SYMBOL	DESCRIPTION
ND, NCNSM1, NB \emptyset X	n	Total number of boxes on all lifting surfaces (panels) = size of the square matrix $[A^{WW}] = [D]$ = matrix of downwash factors.
MD	m	Total number of right hand sides in the system of simultaneous linear equations solved in subroutine $S\emptyset LVIT$.
KD		Working array size (real-variable dimension); presently set to 8000
NI		NTP4; tape number assigned to tape containing all rows of the augmented matrix $[A^{WW} W]$ - see 2.5.1 or 2.5.14
NM		NTP1; tape number, used as scratch unit
N \emptyset		NTP9; tape number, used as scratch unit
NW		NTP3; tape number assigned to tape containing all solutions of the system of simultaneous linear equations
NPR1		Print flag for solutions - input

For present assignment of tape units see 2.4.

2. 5. 17 SUBROUTINE QUAS (ND, MD, KD, NI, MM, NO, NAT, NW,
LTAPE, RHSTAP, NPRI)

Functional Description

Subroutine QUAS is used in conjunction with subroutine FUTSØL for cases with NCØRE ≥ 4000 , and/or N5 = 1. It reads the matrix of downwash factors, [D] from tape NTP1 in row order, and as many columns of right hand sides (m, say), from tape RHSTAP, as the dimension of the work area in QUAS (3700 complex words) will permit. Subroutine QUAS computes the so called quasi-inverse of the matrix [D] and saves it on tape LTAPE; it also computes the solutions for the m right hand sides and saves these on tape NTP3.

Data from Cards

None

Data from Common

None

Data from Tapes

MNEMONICS	SYMBOL	TAPE	DESCRIPTION
A(3700)	$[D],$ $[A^{ww}]$	NTP1	Matrix of downwash factors; see 2.5.8
NMD		NTP4	Total number of modes for case
A(3700)	$[W]$	NTP4	Matrix of right hand sides; see 2.5.5 or 2.5.15

Data to Common

None

Data to Tapes

MNEMONICS	SYMBOL	TAPE	DESCRIPTION
A(3700)		LTAPE	Quasi-inverse matrix of [D]. Master tape has to be specified to preserve quasi-inverse for future analysis; otherwise tape is considered a scratch unit.
A(3700)		NTP3	Solution matrix; dimension $NB\emptyset X \times n$, where n is the number of right hand sides for which solutions were obtained.

Note that Subroutine QUAS uses a single complex work array, A(3700) for all input and output variables.

Data in Argument List

MNEMONICS	SYMBOL	DESCRIPTION
ND, NCNSM1, NB \emptyset X	n	Total number of boxes on all lifting surfaces
MD	m	On output MD = n = the number of right hand sides for which solutions were obtained
KD		Working array size (complex variable dimension); presently set to 3700.
NI		NTP1; tape number assigned to tape containing all rows of matrix [A ^{ww}] (matrix of downwash factors)
MM		NTP8
N \emptyset		NTP9
NAT		NTP2
		} tape numbers for scratch units

MNEMONICS	SYMBOL	DESCRIPTION
NW		NTP3, tape number for the solution matrix
LTAPE		Tape number assigned to the tape containing the quasi-inverse of [D]
RHSTAP		NTP4; tape number assigned to tape containing the matrix of right hand sides

For present assignment of tape units see Sec. 2.4

2.5.18 SUBROUTINE FUTSØL (ND, MD, KD, NL, MM, NØ, NAT, NW, LTAPE, RHSTAP)

Functional Description

Subroutine FUTSØL is used for cases with $NCØRE \geq 4000$, and whenever new modes only are required for a previously computed case ($N1 = 1$). When $N1 = 0$, subroutine FUTSØL is called following subroutine QUAS, and uses the quasi-inverse of the matrix [D] from tape LTAPE. Additional right hand sides are read from tape RHSTAP, and the additional solutions are saved on tape NTP3 following the previously obtained solutions.

Subroutine FUTSØL is bypassed when $N5 = 1$ and $NCØRE < 3700$.

Data from Cards

None

Data from Common

None

Data from Tapes

MNEMONICS	SYMBOL	TAPE	DESCRIPTION
A(3700)	[W]	RHSTAP	Additional columns of right hand sides
A(3700)		LTAPE	The quasi-inverse matrix generated in QUAS

Data to Common

None

Data to Tapes

MNEMONICS	SYMBOL	TAPE	DESCRIPTION
A(3700)		NW	The additional solutions for the additional right hand sides

For present assignment of tape units see Sec. 2.4.

Data in Argument List

Same as for QUAS - see 2.5.17, except that NI is now merely a scratch unit and does not contain the input matrix [D].

2.5.19 SUBROUTINE AIC(BR,S,KR,NBF,NBL,JØBNØ,ICØD,VEL,NGUST,NPR2)

Functional Description

Subroutine AIC is used for AIC-type cases only. It reads the solutions to the AIC-type elastic modes from tape NTP3 and the partitioned integration matrix, [B], from tape NTP10. Using these arrays subroutine AIC supercolumns according to Eq. 2.5-1*, and saves them on tape NTP9.

*See Part I, Vol I

Data from Cards

None

Data from Common

Contents of the labeled common blocks /NTPS/, /XYZ/ and /YZY/.

Data from Tapes

MNEMONICS	SYMBOL	TAPE	DESCRIPTION
NSTRIP	}	NTP10	See 2. 5. 3
NSIZE(50, 2)			
LIM(50, 3)			
B(10, 30)	$[B]_{AIC}$	NTP3	Integration matrix
DIJMIW(400, 4)	$[D]^{-1}[W]_{AIC}$		Solution matrix for the AIC type right hand sides
CPGUST(400)		NTP3	Solution for the gust right hand side - read only if NGUST \neq 0.

Data to Common

None

Data to Tapes

MNEMONICS	SYMBOL	TAPE	DESCRIPTION
CH(4, 4, 50)		NTP9	AIC matrix written one supercolumn at a time; see Eq. 2. 6-2. *

*See Part I, Vol I

Data in Argument List

MNEMONICS	SYMBOL	DESCRIPTION
BK	$b_r, c/2$	Reference semichord
S, REFSPN	$b/2, s$	Reference semispan - input
KR	$b/2$	Reduced frequency
NBF		First bay number
NBL		Last bay number
JØBNØ		AIC-case-identifier
ICØD		1, built in constant
VEL		Aircraft velocity
NGUST		Gust flag
NPR2		Print flag for AIC's

for which
AIC's are
written on
tape

- input

2. 5. 20 SUBROUTINE AERØ (BR, B2, KR, ACAP, NRF, NSV, NYAW,
IMX, NPC, NDELT, NPR3)

Functional Description

Subroutine AERØ is used for AIC-type cases only. It reads the AIC supercolumns from tape NTP9 and computes the stability derivatives for the rigid body modes according to Eqs. (2.4-6) - (2.4-12)*. This subroutine is primarily used for the monitoring of the AIC's. It can be bypassed by setting the data flag NPR3 to 0.

Data from Cards

None

Data from Common

Contents of the labeled commons /NTPS/, /XYZ/ and /ZYZ/.

*See Part I, Vol I

Data from Tapes

MNEMONICS	SYMBOL	TAPE	DESCRIPTION
NSTRIP		NTP10	See 2.5.3
NSIZB(50, 2)			
LIM(50, 3)			
XISLCT(50, 7)			
CH(4, 4, 50)		NTP9	See 2.5.19

Data to Common

None

Data to Tapes

Usually none. However, in special cases, when dynamic stability derivatives are also required (input flag NPR3=3), tape unit 2 is used for saving the total lift-and moment coefficients computed for the steady case ($KR(1) = 0.0$). Subsequently, this information is read from tape unit 2 and is used in the calculation of the dynamic stability derivatives for the unsteady case ($KR(2) > 0.0$) - see 2.4.2, in Part I, Vol I.

Data in Argument List

MNEMONICS	SYMBOL	DESCRIPTION
BR	$b_r, c/2$	See 2.5.19
B2	b_2, s	
KR	k_r	
ACAP, REFARA	A, S	Reference area - input

MNEMONICS	SYMBOL	DESCRIPTION
NRF		Number of reduced frequencies for case - input
NSV		Number of strips on vertical panels (fin) of T-tail cases; 0 otherwise - input
NYAW		Yaw flag - input
IMX		Number of rigid body modes for AIC-type case-from GENB
NPC		Camber flag
NDELT	δ	Symmetry flag
NPR3		Print flag

2. 5. 21 SUBROUTINE FINAL (BR, B2, ACAP, NSV, NBV, IMX, NYAW, NPC, NDELT, FGAMMA)

Functional Description

Subroutine FINAL is used for AIC-type cases only. It reads the solutions for the AIC-type right hand sides from tape NTP3 and computes the pressures for the rigid body modes for all boxes of all lifting surfaces. These are printed along with the box number and the fractional chordwise location of the load points (1/4-chord of box). Subroutine FINAL can be bypassed by setting the data flag NPR3 to 0.

Data from Cards

None

Data from Common

Contents of the labeled commons /NTPS/, /XXZ/, and /YZY/.

Data from Tapes

MNEMONICS	SYMBOL	TAPE	DESCRIPTION
NSTRIP		NTP10	See 2.5.3
NSIZB(50, 2)			
LIM(50, 3)			
CPM(400, 4)		NTP3	Solution matrix for the AIC-type right hand sides

Data to Common

None

Data to Tapes

None

Data in Argument List

MNEMONICS	SYMBOL	DESCRIPTION
BR	b_r	See 2.5.20
B2	$b/2, s$	
ACAP, REFARA	A, S	
NSV		Total number of boxes on vertical panels (fin) of T-tail cases - input
NBV		
IMX		See 2.1.2
NYAW		
NPC		
NDELT	δ	Array of dihedral angles for all strips of all lifting surfaces.
FGAMMA(50)		

2. 5. 22

SUBROUTINE GENF (NBØX, NTØT, NMD, NDELT, NST,
LIM, FL, XCAP, XII, EV, SDELX,
BZ, Y, ZZ, XØC, NBV)

Functional Description

Subroutine GENF is used for modal type cases only. It reads the solutions for all modes from tape NTP3 and writes the pressures for all modes and all boxes along with the x, y, z coordinates of the boxes. It also computes the sectional lift and moment coefficients for all strips of all lifting surfaces as well as the total lift- and moment coefficients - see Eqs. (2.4-6)-(2.4-12)* Subroutine GENF also computes the generalized forces for all pressure modes and deflection modes; see Eq. 2.4-5*

Data from Cards

None

Data from Common

Contents of the labeled common blocks /NTPS/, /XYZ/ and /BØDY/.

Data from Tapes

MNEMONICS	SYMBOL	TAPE	DESCRIPTION
P(400)	$\left\{ \Delta C_p \right\}$	NTP3	Solution matrix for modal type data read in column order. Dimension of matrix is NBØXxNMD.
P(400)	$\left\{ \Delta C_p^{(f)} \right\}$	NTP7	Matrix of pressures for all body elements and modes read in column order. Dimension of matrix is NBExNMD. This data is read for body cases only.
B(400)		NTP10	Integration matrix for modal type data read in column order. Dimension of matrix is NTØTxNMD where NTØT = NBØX + NBE.

Data to Common

None

*See Part I, Vol I

Data to Tapes

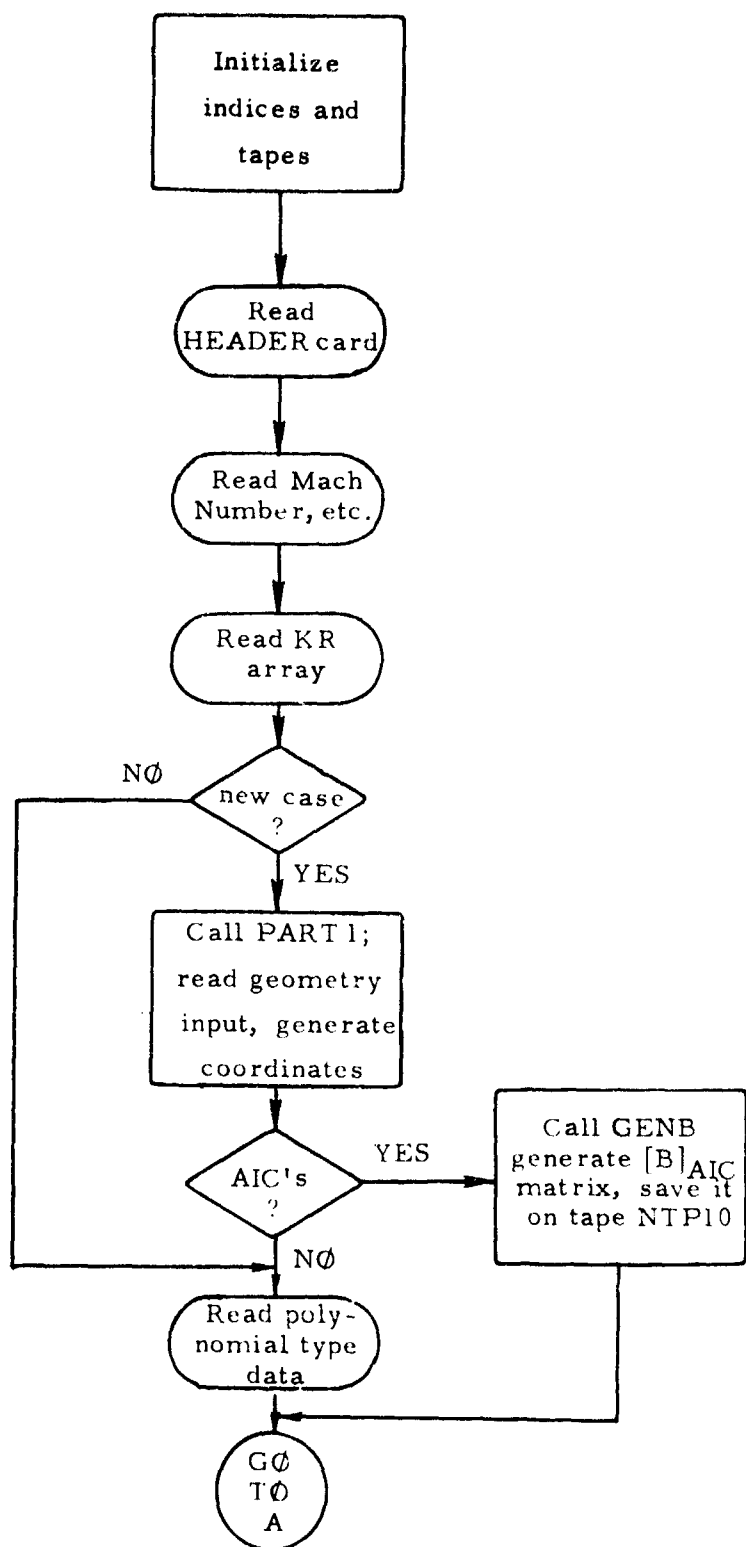
None

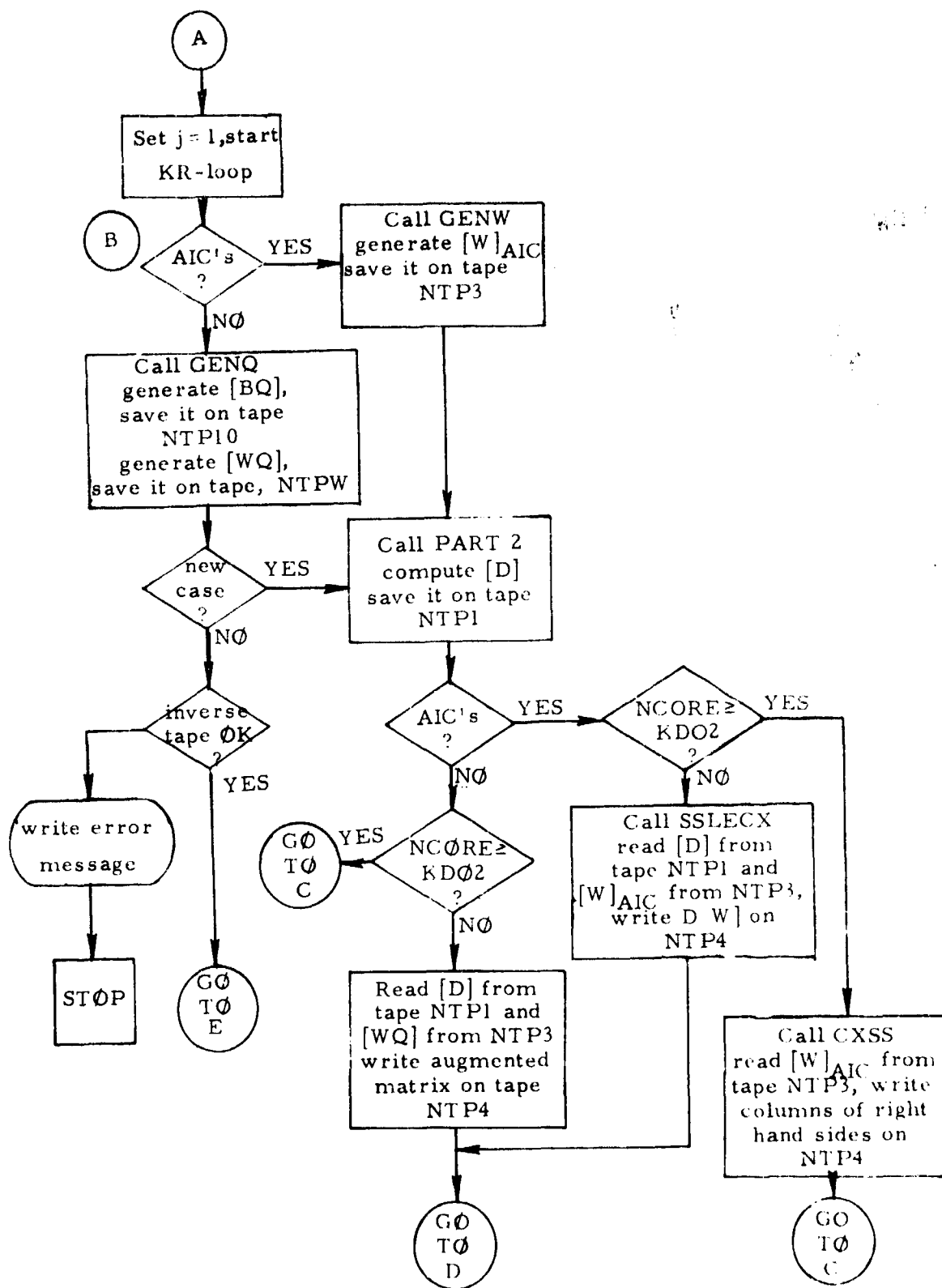
Data in Argument List

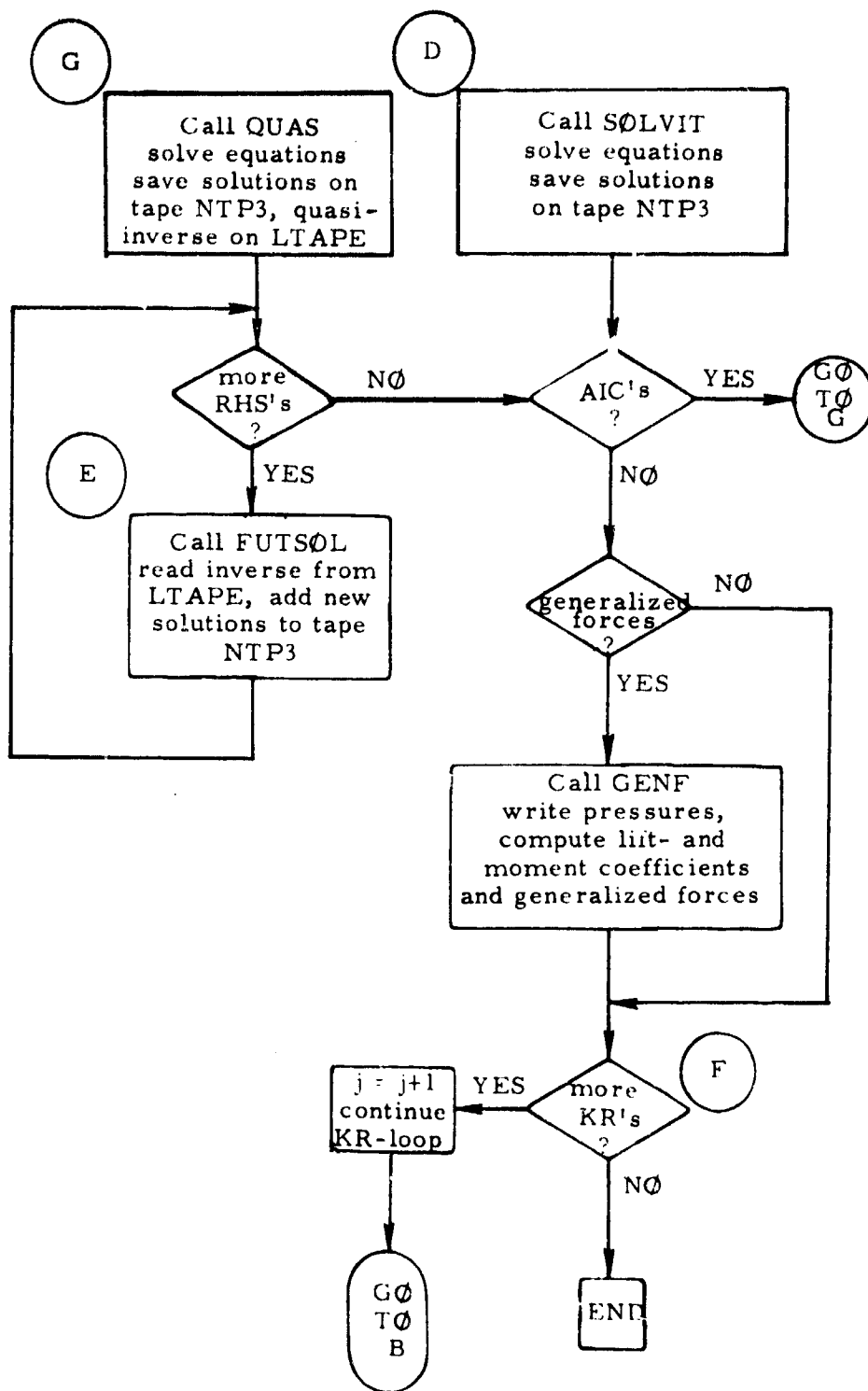
MNEMONICS	SYMBOL	DESCRIPTION
NDELT NB NSTRIP	δ	Symmetry flag Number of bodies } input Number of strips }
NBØX		Total number of boxes on all panels
NTØT	n	NBØX + NBE, where NBE = total number of body elements
NBV		T-tail input
NSV		
NMD	NM, m	Number of modes - input
LIM(50, 3)		See Sec. 2. 5. 3
ACAP, REFARA	A, S	
FL, REFCHD	\bar{c} , $2b_r$	
B2, REFSPN	s, b/2	
EV(400)	ξ	See Blank Common, Sec. 2. 3. 1
Y(400)	y	
ZZ(400)	z	
SDELX(400)	Δx_s	See Common /XXZ/, Sec. 2. 3. 2
XIJ(50)	x_{le}	
XØC(400)	ξ/c	

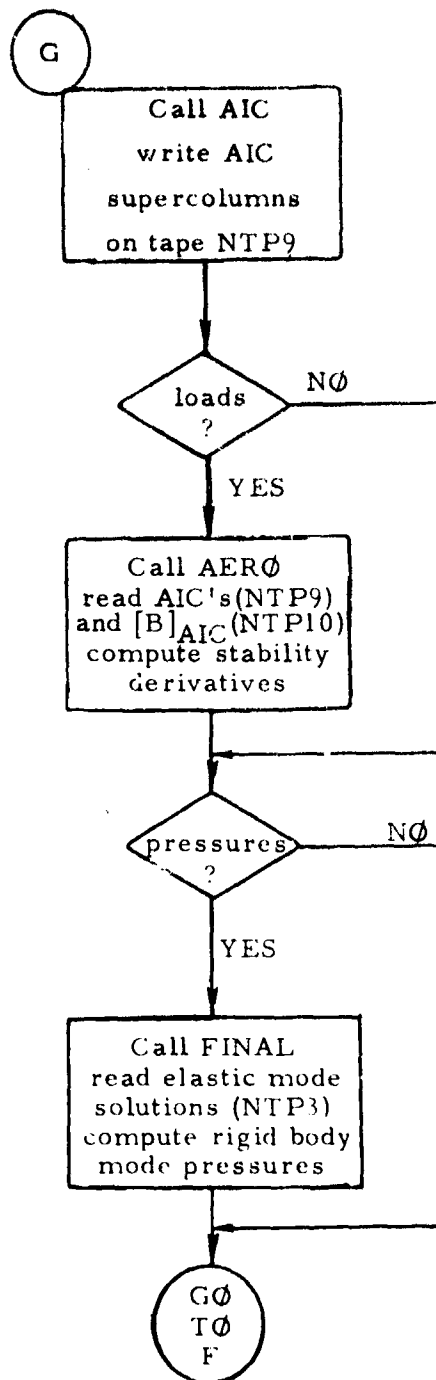
2.6 BLOCK DIAGRAM

The basic block diagram for the computer program H7WC is shown on the next four pages. All logical steps in the "main" part of the program are indicated, and the basic functions of all subroutines that are called from "main" are outlined. A complete description of all subroutines - including those that are not called directly from "main," and therefore are not indicated in the block diagram - can be found in Sections 2.5.1 through 2.5.22.









2.7 PROGRAM LISTING

The following pages (pp. 95 - 143) contain the complete listing of program H7WC. The individual subroutines of the program (including the "main" part of it) are listed in the order of their description in Section 2.5.

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MAIN - EFM SOURCE STATEMENT - (FMS) -

```

NGUST = 0
PI = 3.14159265
KD = 8000
DO 109 K=1,0
  IF (NR(11),EQ,STOP) STOP
  IF (NR(11),EQ,STOP) STOP
  WRITE (6,16) (NR(11),I=1,15)
  READ (5,15) FRACHACAP,FL,62,NOELT,NP,NB,NMF,NCORE,NI,N2,N3,N4
  N = 15/2.0
  NGATA = NS
  NGAPN = NP
  BETA = SORT(LQ-FRACHACAP,2)
  READ (5,23) (AFREQ(I),I=1,NMF)
  WRITE (6,16) (AFREQ(I),I=1,NMF)
  IF (NI,EQ,1) GO TO 110
  CALL PARTIJCUM,COEFF,COEFB,YIN,ZIN)
  NBOZ = NCNMI
  NSTRIP = JCUM
  IF (NB,NE,0) NBE=NBEALNB)
  NTOT = NBOZ+NSE
  READ (5,55) JSPEC,MP,MSV,NBY,NBYO,VEL,VC
  IF (NGUST,NE,0) JSPEC=0.0
  READ (5,47) (LIMIT(I),I=1,31),I=1,NSTRIP:
  110 CONTINUE
  IF (INZ,EQ,0) GO TO 122
  CALL GENDEV,SOELX,DELY,X,NSTRIP,BETA,INX,MP,INP,NSARAY,ALTR
  NBE = NBI
  NSE = NSE
  NSTR = NSE-NBI+1
  NSTRPG=NSTR
  IF (NGUST,NE,0) NSTRPG=NSTR+1
  122 CONTINUE
  NCSF=0
  IF (INZ,EQ,0) REJDS,35) NMU,NTA,N5,N6,N7,NTP1,NTPH,NMTP,NMTB
  IF (IN,EQ,1) WRITE (LTAPE) NTP1
  JRF = J
  IO = 0
  IF (IN,EQ,1) REJDS,35) NMU,NTA,N5,N6,N7,NTP1,NTPH,NMTP,NMTB
  KROBR = REJDS,35) NMU,NTA,N5,N6,N7,NTP1,NTPH,NMTP,NMTB
  IF (INZ,EQ,0) GO TO 125
  CALL GENKROBR,NCSF,TO,MP,NCUST,NG,62AD,62RD)
  GO TO 127
  125 CONTINUE
  CALL GENQIMA,N5,N6,NMF,JRF,NMF,NMO,NMTP,NMTB,NCORE,
  1 CALL K002,YIN,ZIN,COEFB,COEFB1
  IF (NI,EQ,1) GO TO 152
  127 CONTINUE
  RENJMO NTP1
  NBOZ = NB

```

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MAIN - EFM SOURCE STATEMENT - (FMS) -

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HAI0025
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HAI0040
HAI0045
HAI0050
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HAI0060
HAI0065
HAI0070
HAI0075
HAI0080
HAI0085
HAI0090
HAI0095
HAI0100
HAI0105
HAI0110
HAI0115
HAI0120
HAI0125
HAI0130
HAI0135
HAI0140
HAI0145
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HAI0175
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HAI0200
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HAI0240
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HAI0250
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HAI0265
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HAI0280
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HAI0300
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HAI0345
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HAI0355
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HAI0365
HAI0370
HAI0375
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01/15/71

MAIN - EFM SOURCE STATEMENT - (FMS) -

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HAI0090
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HAI0550

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GENM - EFM SOURCE STATEMENT - IFN(5) -

```

X5 = XLE + RH(1,5) * CHORD
X6 = XLE + RH(1,6) * CHORD
X7 = XLE + RH(1,7) * CHORD
IF (MPC-NE-0) GO TO 290
GO TO 130,150,170,190,210,230,460,4601, J
130 DO 140 L = LIM1,LIM2
IR = L - LIM1 + 1
140 WIR,IC1 = CMPLX(0.0,KROBR)
GO TO 440
150 DO 160 L = LIM1,LIM2
IR = L - LIM1 + 1
160 WIR,IC1 = CMPLX(1.0,KROBR*(XIL1-X1))
GO TO 440
170 DO 180 L = LIM1,LIM2
IR = L - LIM1 + 1
180 WIR,IC1 = CMPLX(1.0,KROBR*(XIL1-X1))
GO TO 440
190 DO 200 L = LIM1,LIM2
IR = L - LIM1 + 1
200 WIR,IC1 = CMPLX(1.0,KROBR*(XIL1-X5))
GO TO 440
210 DO 220 L = LIM1,LIM2
IR = L - LIM1 + 1
220 WIR,IC1 = CMPLX(1.0,KROBR)
GO TO 440
230 DO 240 L = LIM1,LIM2
IR = L - LIM1 + 1
240 WIR,IC1 = CMPLX(1.0,KROBR)
GO TO 440
250 CONTINUE
GO TO 1240,300,340,380,400,420,440,4601, J
260 DO 290 L = LIM1,LIM2
IR = L - LIM1 + 1
290 TOP1 = (XIL1-X2) * (XIL1-X3)
TOP1 = (XIL1-X2) * (XIL1-X3)

```

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GENM - EFM SOURCE STATEMENT - IFN(5) -

```

BOT = BOT1 + 1
IF (XISCT(1,4),HE-0.0,AND,XIL1-GT,X4) GO TO 270
WIR = TOPR/BO1
WIM = KROBR*(TOP1/BO1)
GO TO 280
270 D = ((X5-X2) * (X5-X3))/BOT
DELTA = ((X5-X2) * (X5-X3))/BOT
WIR = KROBR*(D + DELTA * (XIL1-X5))
WIM = DELTA
280 WIR,IC1 = CMPLX(WIR,WIM)
290 CONTINUE
GO TO 480
300 DO 310 L = LIM1,LIM2
IR = L - LIM1 + 1
310 TOP1 = (XIL1-X1) * (XIL1-X3)
TOP1 = (XIL1-X1) * (XIL1-X3)
BOT = BOT2
IF (XISCT(1,4),HE-0.0,AND,XIL1-GT,X4) GO TO 310
WIR = TOPR/BO1
WIM = KROBR*(TOP1/BO1)
GO TO 320
320 D = ((X5-X1) * (X5-X3))/BOT
DELTA = ((X5-X1) * (X5-X3))/BOT
WIR = DELTA
WIM = KROBR*(D + DELTA*(XIL1-X5))
330 WIR,IC1 = CMPLX(WIR,WIM)
340 DO 350 L = LIM1,LIM2
IR = L - LIM1 + 1
350 TOP1 = (XIL1-X1) * (XIL1-X2)
TOP1 = (XIL1-X1) * (XIL1-X2)
BOT = BOT3
IF (XISCT(1,4),HE-0.0,AND,XIL1-GT,X4) GO TO 350
WIR = TOPR/BO1
WIM = KROBR*(TOP1/BO1)
GO TO 360
360 D = ((X5-X1) * (X5-X2))/BOT
DELTA = ((X5-X1) * (X5-X2))/BOT
WIR = DELTA
WIM = KROBR*(D + DELTA*(XIL1-X5))
370 WIR,IC1 = CMPLX(WIR,WIM)
380 DO 390 L = LIM1,LIM2
IR = L - LIM1 + 1
390 TOP1 = (XIL1-X1) * (XIL1-X5)
TOP1 = (XIL1-X1) * (XIL1-X5)
WIR,IC1 = CMPLX(WIR,WIM)
390 CONTINUE
GO TO 400
400 DO 410 L = LIM1,LIM2
IR = L - LIM1 + 1
410 TOP1 = (XIL1-X1) * (XIL1-X5)
TOP1 = (XIL1-X1) * (XIL1-X5)
WIR,IC1 = CMPLX(WIR,WIM)
410 CONTINUE
GO TO 440

```



```

01/15/71
GENO - EPM SOURCE STATEMENT - (PMSI) -
10 IF (PMSI.EQ.0) GO TO 11
11 CONTINUE
12 IF (PMSI.EQ.1) GO TO 13
13 CONTINUE
14 IF (PMSI.EQ.2) GO TO 15
15 CONTINUE
16 IF (PMSI.EQ.3) GO TO 17
17 CONTINUE
18 IF (PMSI.EQ.4) GO TO 19
19 CONTINUE
20 IF (PMSI.EQ.5) GO TO 21
21 CONTINUE
22 IF (PMSI.EQ.6) GO TO 23
23 CONTINUE
24 IF (PMSI.EQ.7) GO TO 25
25 CONTINUE
26 IF (PMSI.EQ.8) GO TO 27
27 CONTINUE
28 IF (PMSI.EQ.9) GO TO 29
29 CONTINUE
30 IF (PMSI.EQ.10) GO TO 31
31 CONTINUE
32 IF (PMSI.EQ.11) GO TO 33
33 CONTINUE
34 IF (PMSI.EQ.12) GO TO 35
35 CONTINUE
36 IF (PMSI.EQ.13) GO TO 37
37 CONTINUE
38 IF (PMSI.EQ.14) GO TO 39
39 CONTINUE
40 IF (PMSI.EQ.15) GO TO 41
41 CONTINUE
42 IF (PMSI.EQ.16) GO TO 43
43 CONTINUE
44 IF (PMSI.EQ.17) GO TO 45
45 CONTINUE
46 IF (PMSI.EQ.18) GO TO 47
47 CONTINUE
48 IF (PMSI.EQ.19) GO TO 49
49 CONTINUE
50 IF (PMSI.EQ.20) GO TO 51
51 CONTINUE
52 IF (PMSI.EQ.21) GO TO 53
53 CONTINUE
54 IF (PMSI.EQ.22) GO TO 55
55 CONTINUE
56 IF (PMSI.EQ.23) GO TO 57
57 CONTINUE
58 IF (PMSI.EQ.24) GO TO 59
59 CONTINUE
60 IF (PMSI.EQ.25) GO TO 61
61 CONTINUE
62 IF (PMSI.EQ.26) GO TO 63
63 CONTINUE
64 IF (PMSI.EQ.27) GO TO 65
65 CONTINUE
66 IF (PMSI.EQ.28) GO TO 67
67 CONTINUE
68 IF (PMSI.EQ.29) GO TO 69
69 CONTINUE
69 CONTINUE

```

```

01/15/71
GENO - EPM SOURCE STATEMENT - (PMSI) -
10 IF (PMSI.EQ.0) GO TO 11
11 CONTINUE
12 IF (PMSI.EQ.1) GO TO 13
13 CONTINUE
14 IF (PMSI.EQ.2) GO TO 15
15 CONTINUE
16 IF (PMSI.EQ.3) GO TO 17
17 CONTINUE
18 IF (PMSI.EQ.4) GO TO 19
19 CONTINUE
20 IF (PMSI.EQ.5) GO TO 21
21 CONTINUE
22 IF (PMSI.EQ.6) GO TO 23
23 CONTINUE
24 IF (PMSI.EQ.7) GO TO 25
25 CONTINUE
26 IF (PMSI.EQ.8) GO TO 27
27 CONTINUE
28 IF (PMSI.EQ.9) GO TO 29
29 CONTINUE
30 IF (PMSI.EQ.10) GO TO 31
31 CONTINUE
32 IF (PMSI.EQ.11) GO TO 33
33 CONTINUE
34 IF (PMSI.EQ.12) GO TO 35
35 CONTINUE
36 IF (PMSI.EQ.13) GO TO 37
37 CONTINUE
38 IF (PMSI.EQ.14) GO TO 39
39 CONTINUE
40 IF (PMSI.EQ.15) GO TO 41
41 CONTINUE
42 IF (PMSI.EQ.16) GO TO 43
43 CONTINUE
44 IF (PMSI.EQ.17) GO TO 45
45 CONTINUE
46 IF (PMSI.EQ.18) GO TO 47
47 CONTINUE
48 IF (PMSI.EQ.19) GO TO 49
49 CONTINUE
50 IF (PMSI.EQ.20) GO TO 51
51 CONTINUE
52 IF (PMSI.EQ.21) GO TO 53
53 CONTINUE
54 IF (PMSI.EQ.22) GO TO 55
55 CONTINUE
56 IF (PMSI.EQ.23) GO TO 57
57 CONTINUE
58 IF (PMSI.EQ.24) GO TO 59
59 CONTINUE
60 IF (PMSI.EQ.25) GO TO 61
61 CONTINUE
62 IF (PMSI.EQ.26) GO TO 63
63 CONTINUE
64 IF (PMSI.EQ.27) GO TO 65
65 CONTINUE
66 IF (PMSI.EQ.28) GO TO 67
67 CONTINUE
68 IF (PMSI.EQ.29) GO TO 69
69 CONTINUE
69 CONTINUE

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01/15/71

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100 CONTINUE
110 CONTINUE
120 CONTINUE
130 CONTINUE
140 CONTINUE
150 CONTINUE
160 CONTINUE
170 CONTINUE
180 CONTINUE
190 CONTINUE
200 CONTINUE
210 CONTINUE
220 CONTINUE
230 CONTINUE
240 CONTINUE
250 CONTINUE
260 CONTINUE
270 CONTINUE
280 CONTINUE
290 CONTINUE
300 CONTINUE
310 CONTINUE
320 CONTINUE
330 CONTINUE
340 CONTINUE
350 CONTINUE
360 CONTINUE
370 CONTINUE
380 CONTINUE
390 CONTINUE
400 CONTINUE
410 CONTINUE
420 CONTINUE
430 CONTINUE
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450 CONTINUE
460 CONTINUE
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480 CONTINUE
490 CONTINUE
500 CONTINUE
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520 CONTINUE
530 CONTINUE
540 CONTINUE
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610 CONTINUE
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680 CONTINUE
690 CONTINUE
700 CONTINUE
710 CONTINUE
720 CONTINUE
730 CONTINUE
740 CONTINUE
750 CONTINUE
760 CONTINUE
770 CONTINUE
780 CONTINUE
790 CONTINUE
800 CONTINUE
810 CONTINUE
820 CONTINUE
830 CONTINUE
840 CONTINUE
850 CONTINUE
860 CONTINUE
870 CONTINUE
880 CONTINUE
890 CONTINUE
900 CONTINUE
910 CONTINUE
920 CONTINUE
930 CONTINUE
940 CONTINUE
950 CONTINUE
960 CONTINUE
970 CONTINUE
980 CONTINUE
990 CONTINUE
1000 CONTINUE

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TEER - EFM SOURCE STATEMENT - (PMS)

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1. The first part of the document is a list of names and addresses. The names are listed in the first column, and the addresses are listed in the second column. The names are: John Doe, Jane Smith, and Bob Johnson. The addresses are: 123 Main Street, 456 Elm Street, and 789 Oak Street.

2. The second part of the document is a list of names and addresses. The names are listed in the first column, and the addresses are listed in the second column. The names are: John Doe, Jane Smith, and Bob Johnson. The addresses are: 123 Main Street, 456 Elm Street, and 789 Oak Street.

3. The third part of the document is a list of names and addresses. The names are listed in the first column, and the addresses are listed in the second column. The names are: John Doe, Jane Smith, and Bob Johnson. The addresses are: 123 Main Street, 456 Elm Street, and 789 Oak Street.

4. The fourth part of the document is a list of names and addresses. The names are listed in the first column, and the addresses are listed in the second column. The names are: John Doe, Jane Smith, and Bob Johnson. The addresses are: 123 Main Street, 456 Elm Street, and 789 Oak Street.

5. The fifth part of the document is a list of names and addresses. The names are listed in the first column, and the addresses are listed in the second column. The names are: John Doe, Jane Smith, and Bob Johnson. The addresses are: 123 Main Street, 456 Elm Street, and 789 Oak Street.

6. The sixth part of the document is a list of names and addresses. The names are listed in the first column, and the addresses are listed in the second column. The names are: John Doe, Jane Smith, and Bob Johnson. The addresses are: 123 Main Street, 456 Elm Street, and 789 Oak Street.

7. The seventh part of the document is a list of names and addresses. The names are listed in the first column, and the addresses are listed in the second column. The names are: John Doe, Jane Smith, and Bob Johnson. The addresses are: 123 Main Street, 456 Elm Street, and 789 Oak Street.

8. The eighth part of the document is a list of names and addresses. The names are listed in the first column, and the addresses are listed in the second column. The names are: John Doe, Jane Smith, and Bob Johnson. The addresses are: 123 Main Street, 456 Elm Street, and 789 Oak Street.

9. The ninth part of the document is a list of names and addresses. The names are listed in the first column, and the addresses are listed in the second column. The names are: John Doe, Jane Smith, and Bob Johnson. The addresses are: 123 Main Street, 456 Elm Street, and 789 Oak Street.

10. The tenth part of the document is a list of names and addresses. The names are listed in the first column, and the addresses are listed in the second column. The names are: John Doe, Jane Smith, and Bob Johnson. The addresses are: 123 Main Street, 456 Elm Street, and 789 Oak Street.

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SOURCE STATEMENT - IP4451 -

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<pre> SSLECX - EFN SOURCE STATEMENT - IFN(5) - 120 CONTINUE IC2 = IC2 + NSIZE(1,1) 130 CONTINUE IC1 = IC2 - NSIZE(2,1) + 1 DO 130 I = 1, IC1 UR(1) = REAL (W(1),IC1) UR(1) = APMAG (W(1),IC1) DO 140 I = 1, NS1,NS2 IN = I - 1 IN = I - 1 - CPMX(UR(1),W(1)) 140 CONTINUE GO TO 180 150 READ (NTP3) W NS1 = 1 NS2 = NSIZE(1,1) DO 160 I = 1, NS1 UR(1) = REAL(W(1),1) DO 170 I = 1, NS1,NS2 IN = I - 1 IN = I - 1 - CPMX(UR(1),W(1)) 170 CONTINUE IF (NGUST-NE.0) AUGMH(NP2I) = GUST(I) IN = IN + 1 INP1 = IN + 1 180 WRITE (NTP4) AUGMH RETURN END </pre>	<pre> SUBROUTINE C555 (NCKNRI,MD,NGUST) COMMON /NTP5/ NTP1,NTP2,NTP3,NTP4,NTP5,NTP6,NTP7,NTP8,NTP9,NTP10,C5550015 DIMENSION NSIZE(50,2), L1(50,3) COMPLEX (100,354,100) CUST(400), W(50,10) 10 FORMAT (100,354,100COLUMNS NO.,10,27H OF THE AIC-TYPE RMS-MATRIX/ 20 NMOD = (NTP1) NPI = 0 REMIND NTP3 REMIND NTP4 REMIND NTP5 REMIND NTP6 REMIND NTP7 REMIND NTP8 REMIND NTP9 REMIND NTP10 NSTRIP,NSIZE,L1 NMOD = 0 NDRUM = 0 DO 50 K=1,NSTRIP 50 MD = MD+NSIZE(K,1) IF (NGUST-EO.0) GO TO 60 IF (NTP1) NSTRIP,NSIZE,L1 READ (NTP3) CUST 60 CONTINUE WRITE (NTP4) MD DO 300 K=1,NSTRIP NMOD = NMOD+1 NPI = NPI+1 L1 = L1(K,2) L2 = L1(K,2) DO 200 J=1,NMOD NDRUM = NDRUM+1 DO 100 I=1,NSIZE(K,1) W(1) = CUST(I) DO 150 L=1,L2 L1 = L1+1 W(1) = W(1)+L1 150 CONTINUE IF (NPI-NE.0) GO TO 180 WRITE (NTP4) NMOD 160 CONTINUE 170 CONTINUE 180 CONTINUE 190 CONTINUE 200 CONTINUE 300 CONTINUE RETURN END </pre>

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SOLVIT - EFM SOURCE STATEMENT - IFN(5) -
C SUM = 0.0
C SUMM = 0.0
C SUMT = 0.0
C
C N2 = NP + 16
C DO 120 I = 1, 10
C   NP = NT + 10
C   NP = NP + N2 - 10
C 120 SUM = SUM + AIN(I) * AIN(I)
C
C SUM = SUM + REAL(AIN(I)) * REAL(AIN(I)) - AIN(AIN(I)) *
C   AIN(AIN(I))
C 130 SUMT = SUMT + REAL(AIN(I)) * AIN(AIN(I)) + AIN(AIN(I)) *
C   REAL(AIN(I))
C 135 AIN(I) = (SUMT - SUM) / AIN(I)
C
C DENOM = REAL(AIN(I)) * 2 + AIN(AIN(I)) * 2
C TEMP1 = REAL(AIN(I)) - SUM
C TEMP2 = AIN(AIN(I)) - SUMT
C REAL(TEMP1) = (TEMP1 * REAL(AIN(I)) + TEMP2 * AIN(AIN(I))) / DENOM
C REAL(TEMP2) = (TEMP2 * REAL(AIN(I)) - TEMP1 * AIN(AIN(I))) / DENOM
C 140 CONTINUE
C -- MOVE THE SOLUTIONS TO CONTIGUOUS LOCATIONS STARTING AT AIN1
C
C N1 = 4096 + 1
C DO 140 M = 1, 10
C   N2 = N1 - 1
C   N3 = N1 - 1
C   N4 = N1 - 1
C 145 AIN(I) = AIN(N1)
C 146 N1 = N1 - 10
C -- WRITE THE SOLUTIONS ON TAPE
C
C WRITE (UNIT, *)
C N5 = N1 - 1
C DO 145 M = 1, 10
C   N6 = N5 - 1
C   N7 = N5 - 1
C 145 WRITE (UNIT, 1) AIN(N1), 10 = NT, N6, N7
C -- FIRST 10 PASSES ARE THE FIRST PASS
C
C 10 CLASS 10 TO 700
C -- THE FIRST 1000 SOLUTIONS ARE TRIANGULAR MATRICES TO REFLECT THE EFFECT OF
C   THE SOLUTIONS OBTAINED IN PASSES 10 TO 711

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SOLVIT - EFM SOURCE STATEMENT - IFN(5) -
C * NOTE--LOCATIONS AIN(1) TO AIN(11) ARE NOW FREE TO USE
C -- CALCULATE THE NEXT VALUES OF "NEL" AND "NREM"
C
C NELD = NEL
C NOLD = N
C NEL = NEL - K
C NREM = NREM - K
C -- NOW APPLY THE INCREIBLE FORMULA FOR THE NEW "N"
C
C K = (-4 * M - 1) / 2 + ((1/5) * (10.25 + FLOAT((M * 2) * M +
C   1 * (NELD - NOLD)))
C NOLD = NELD - K
C IF (K - LT, NREM) GO TO 150
C LAST = -TRUE
C NREM = 1
C K = NREM
C 150 NT = NELD + 1
C -- READ IN THE ROWS TO BE MODIFIED
C
C DO 160 I = 1, NREM
C   NT = NT + 1
C   IF (I - LT, NREM) GO TO 160
C   NT = NT + N
C 160 READ (RT, 1) NN, (A(10), 10 = N5, NT)
C   NP = N1 - 1
C   NP = NP - 1
C   NN = NN - KOLD
C   NN = NN - KOLD
C   DO 170 M = 1, M
C     N2 = NP + M
C     N3 = NP + M
C     N4 = N1
C     SUM = 0.0
C     SUMR = 0.0
C     SUMT = 0.0
C     DO 165 I = 1, NOLD
C       SUM = SUMT + AIN(I) * AIN(I)
C       SUMR = SUMR + REAL(AIN(I)) * REAL(AIN(I)) - AIN(AIN(I)) *
C         AIN(AIN(I))
C       SUMT = SUMT + REAL(AIN(I)) * AIN(AIN(I)) + AIN(AIN(I)) *
C         REAL(AIN(I))
C     N2 = N2 + 1
C     N3 = N3 + 1
C     N4 = N4 + 1
C 170 AIN(I) = AIN(I) - SUM

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SOLVIT - EFM SOURCE STATEMENT - IFN(5) -
C * NOTE--LOCATIONS AIN(1) TO AIN(11) ARE NOW FREE TO USE
C -- CALCULATE THE NEXT VALUES OF "NEL" AND "NREM"
C
C NELD = NEL
C NOLD = N
C NEL = NEL - K
C NREM = NREM - K
C -- NOW APPLY THE INCREIBLE FORMULA FOR THE NEW "N"
C
C K = (-4 * M - 1) / 2 + ((1/5) * (10.25 + FLOAT((M * 2) * M +
C   1 * (NELD - NOLD)))
C NOLD = NELD - K
C IF (K - LT, NREM) GO TO 150
C LAST = -TRUE
C NREM = 1
C K = NREM
C 150 NT = NELD + 1
C -- READ IN THE ROWS TO BE MODIFIED
C
C DO 160 I = 1, NREM
C   NT = NT + 1
C   IF (I - LT, NREM) GO TO 160
C   NT = NT + N
C 160 READ (RT, 1) NN, (A(10), 10 = N5, NT)
C   NP = N1 - 1
C   NP = NP - 1
C   NN = NN - KOLD
C   NN = NN - KOLD
C   DO 170 M = 1, M
C     N2 = NP + M
C     N3 = NP + M
C     N4 = N1
C     SUM = 0.0
C     SUMR = 0.0
C     SUMT = 0.0
C     DO 165 I = 1, NOLD
C       SUM = SUMT + AIN(I) * AIN(I)
C       SUMR = SUMR + REAL(AIN(I)) * REAL(AIN(I)) - AIN(AIN(I)) *
C         AIN(AIN(I))
C       SUMT = SUMT + REAL(AIN(I)) * AIN(AIN(I)) + AIN(AIN(I)) *
C         REAL(AIN(I))
C     N2 = N2 + 1
C     N3 = N3 + 1
C     N4 = N4 + 1
C 170 AIN(I) = AIN(I) - SUM

```


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QUAS - EFM SOURCE STATEMENT - (FMS) -

QUAS (6,2) (AINT), NM = NS, NT1
 230 WRITE (NM, NT1) (AINT, NM)
 C --- IF TAPE HAS NEVER SWITCHED IT WOULD BE FOOLISH TO SWITCH BACK
 IF JPASS1 EQ 0
 IF JPASS1 EQ 0
 C --- SWITCH TAPES
 C --- BACK SO THAT NT WILL CONTAIN THE TRAPEZOIDAL MATRIX
 C --- TAPE WILL HAVE NOTHING USEFUL ON IT.
 NTAPE = NT
 NT = NT
 REMIND NTAPE
 240 IF JPASS1 EQ 0
 250 READ (NTAPE) NT
 KRED = 0
 DO 297 I=1, MCNT
 READ (NTAPE) KRED
 KRED = KRED + KREAD
 304 KRED = KRED + (N - KRED - 1)
 KREAD = KREAD + 1
 DO 297 LREAD=1, KREAD
 297 READ (NTAPE) LREAD
 DO 299 I=1, MCNT
 299 WRITE (NM, NT1) (LREAD, I, LREAD)
 C --- REMIND ALL FILES EXCEPT THE OUTPUT FILE NM
 C
 REMIND NTAPE
 REMIND NM
 REMIND NT
 REMIND NT
 REMIND NT
 IF I RHSTAP .NE. 0 I REMIND RHSTAP
 DO 299 I=1, MCNT
 299 WRITE (NM, NT1) (I, LREAD, I, LREAD)
 C
 9999 RETURN
 END

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QUAS - EFM SOURCE STATEMENT - (FMS) -

QUAS (6,2) (AINT), NM = NS, NT1
 230 WRITE (NM, NT1) (AINT, NM)
 C --- IF TAPE HAS NEVER SWITCHED IT WOULD BE FOOLISH TO SWITCH BACK
 IF JPASS1 EQ 0
 IF JPASS1 EQ 0
 C --- SWITCH TAPES
 C --- BACK SO THAT NT WILL CONTAIN THE TRAPEZOIDAL MATRIX
 C --- TAPE WILL HAVE NOTHING USEFUL ON IT.
 NTAPE = NT
 NT = NT
 REMIND NTAPE
 240 IF JPASS1 EQ 0
 250 READ (NTAPE) NT
 KRED = 0
 DO 297 I=1, MCNT
 READ (NTAPE) KRED
 KRED = KRED + KREAD
 304 KRED = KRED + (N - KRED - 1)
 KREAD = KREAD + 1
 DO 297 LREAD=1, KREAD
 297 READ (NTAPE) LREAD
 DO 299 I=1, MCNT
 299 WRITE (NM, NT1) (LREAD, I, LREAD)
 C --- REMIND ALL FILES EXCEPT THE OUTPUT FILE NM
 C
 REMIND NTAPE
 REMIND NM
 REMIND NT
 REMIND NT
 REMIND NT
 IF I RHSTAP .NE. 0 I REMIND RHSTAP
 DO 299 I=1, MCNT
 299 WRITE (NM, NT1) (I, LREAD, I, LREAD)
 C
 9999 RETURN
 END

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QUAS - EFM SOURCE STATEMENT - (FMS) -

QUAS (6,2) (AINT), NM = NS, NT1
 230 WRITE (NM, NT1) (AINT, NM)
 C --- IF TAPE HAS NEVER SWITCHED IT WOULD BE FOOLISH TO SWITCH BACK
 IF JPASS1 EQ 0
 IF JPASS1 EQ 0
 C --- SWITCH TAPES
 C --- BACK SO THAT NT WILL CONTAIN THE TRAPEZOIDAL MATRIX
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 NTAPE = NT
 NT = NT
 REMIND NTAPE
 240 IF JPASS1 EQ 0
 250 READ (NTAPE) NT
 KRED = 0
 DO 297 I=1, MCNT
 READ (NTAPE) KRED
 KRED = KRED + KREAD
 304 KRED = KRED + (N - KRED - 1)
 KREAD = KREAD + 1
 DO 297 LREAD=1, KREAD
 297 READ (NTAPE) LREAD
 DO 299 I=1, MCNT
 299 WRITE (NM, NT1) (LREAD, I, LREAD)
 C --- REMIND ALL FILES EXCEPT THE OUTPUT FILE NM
 C
 REMIND NTAPE
 REMIND NM
 REMIND NT
 REMIND NT
 REMIND NT
 IF I RHSTAP .NE. 0 I REMIND RHSTAP
 DO 299 I=1, MCNT
 299 WRITE (NM, NT1) (I, LREAD, I, LREAD)
 C
 9999 RETURN
 END

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QUAS - EFM SOURCE STATEMENT - (FMS) -

QUAS (6,2) (AINT), NM = NS, NT1
 230 WRITE (NM, NT1) (AINT, NM)
 C --- IF TAPE HAS NEVER SWITCHED IT WOULD BE FOOLISH TO SWITCH BACK
 IF JPASS1 EQ 0
 IF JPASS1 EQ 0
 C --- SWITCH TAPES
 C --- BACK SO THAT NT WILL CONTAIN THE TRAPEZOIDAL MATRIX
 C --- TAPE WILL HAVE NOTHING USEFUL ON IT.
 NTAPE = NT
 NT = NT
 REMIND NTAPE
 240 IF JPASS1 EQ 0
 250 READ (NTAPE) NT
 KRED = 0
 DO 297 I=1, MCNT
 READ (NTAPE) KRED
 KRED = KRED + KREAD
 304 KRED = KRED + (N - KRED - 1)
 KREAD = KREAD + 1
 DO 297 LREAD=1, KREAD
 297 READ (NTAPE) LREAD
 DO 299 I=1, MCNT
 299 WRITE (NM, NT1) (LREAD, I, LREAD)
 C --- REMIND ALL FILES EXCEPT THE OUTPUT FILE NM
 C
 REMIND NTAPE
 REMIND NM
 REMIND NT
 REMIND NT
 REMIND NT
 IF I RHSTAP .NE. 0 I REMIND RHSTAP
 DO 299 I=1, MCNT
 299 WRITE (NM, NT1) (I, LREAD, I, LREAD)
 C
 9999 RETURN
 END

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AIC - EFN SOURCE STATEMENT - IFMIS) -

CH11JJ.MK3 = CH11JJ.MK1+0.111,1)DIJ.MILA,2)B OF ACTOR

440 CONTINUE (G,0) GO TO 530

IF (GUST(1,1) = 0) GO TO 530

DO 520 I1 = 1,NDOP1

G(I1) = 0.

DO 520 I1 = 1,NDOP2

G(I1) = LING + L

520 CONTINUE (G,0) GO TO 537

530 CONTINUE (G,0) GO TO 537

537 CONTINUE (G,0) GO TO 537

540 CONTINUE (G,0) GO TO 537

END

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AIC - EFN SOURCE STATEMENT - IFMIS) -

CH11JJ.MK3 = CH11JJ.MK1+0.111,1)DIJ.MILA,2)B OF ACTOR

440 CONTINUE (G,0) GO TO 530

IF (GUST(1,1) = 0) GO TO 530

DO 520 I1 = 1,NDOP1

G(I1) = 0.

DO 520 I1 = 1,NDOP2

G(I1) = LING + L

520 CONTINUE (G,0) GO TO 537

530 CONTINUE (G,0) GO TO 537

537 CONTINUE (G,0) GO TO 537

540 CONTINUE (G,0) GO TO 537

END

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AIC - EFN SOURCE STATEMENT - IFMIS) -

CH11JJ.MK3 = CH11JJ.MK1+0.111,1)DIJ.MILA,2)B OF ACTOR

440 CONTINUE (G,0) GO TO 530

IF (GUST(1,1) = 0) GO TO 530

DO 520 I1 = 1,NDOP1

G(I1) = 0.

DO 520 I1 = 1,NDOP2

G(I1) = LING + L

520 CONTINUE (G,0) GO TO 537

530 CONTINUE (G,0) GO TO 537

537 CONTINUE (G,0) GO TO 537

540 CONTINUE (G,0) GO TO 537

END

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13. ABSTRACT <p>Two methods of accounting for body-lifting surface interference in unsteady flow are considered. The first method is described in Part I of this report, while the second will be described in Part II to follow.</p> <p>The first method is a direct application of nonplanar lifting surface elements to both the lifting surfaces and the body surfaces. The body is treated as an annular wing. This type of idealization must be used with an axial doublet introduced to account for body incidence effects. The undesirable effects of the annular wing representation are then reduced.</p> <p>The second approach, to be described in Part II, uses an image system and an axial singularity system to account for the effects of the bodies.</p> <p>This report also describes an improvement of the Doublet-Lattice Method of Albano and Rodden. The improvement pertains to wing-tail problems where there exists a small vertical (non-zero) separation between the wing and tail planes. Such problems can now be handled with ease.</p>		

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